UNITED STATES OF AMERICA DEPARTMENT OF TRANSPORTATION-FEDERAL AVIATION ADMINISTRATION STANDARD AIRWORTHINESS CERTIFICATE

1 NATIONALITY AND REGISTRATION MARKS 2 MANUFACTURER AND MODEL N272BG

Gulfstream GV-SP (G550)

3 AIRCRAFT SERIAL NUMBER

5164

4 CATEGORY

Transport

5 AUTHORITY AND BASIS FOR ISSUANCE

This airworthiness certificate is issued pursuant to 49 U.S.C. § 44704 and certifies that, as of the date of issuance, this aircraft has been inspected and found to conform to its type certificate and be in a condition for safe operation. This aircraft meets the requirements of the applicable airworthiness standards in Annex 8 to the Convention on International Civil Aviation, except as follows:

Exemption 7946 25.813 (e) Door Between Passenger Compartments

Exemption 8004 25.901 (c) Single Failure Criteria Exemption 8142 25.901 (c) Single Failure Criteria

6 TERMS AND CONDITIONS

Unless sooner surrendered, suspended, revoked, or a termination date is otherwise established by the FAA, this airworthiness certificate is effective as long as maintenance, preventative maintenance, and alterations are performed per the applicable Federal Aviation Regulations and the aircraft is registered in the United States.

DATE OF ISSUANCE R 21 Aug 2007 FAA REPRESENTATIVE MICHAEL Edward Masterson Ma

Date: 2018.09.17 14:52:56 -04'00'

DESIGNATION NUMBER FSDO EA-17

Any alteration, misuse, or reproduction of this certificate for a fraudulent purpose may be punishable by certificate revocation, fine, and / or imprisonment.

THIS CERTIFICATE MUST BE DISPLAYED IN THE AIRCRAFT PER THE APPLICABLE FEDERAL AVIATION REGULATIONS.

FAA Form 8100-2 (11-2016) Previous Edition Dated 04-11 May be Used Until Depleted



DEPARTMENT OF THANSPORTATION—I JOERAL AVIATION ADMINISTRATION

STANDARD AIRWORTHINESS CERTIFICATE

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1.	NATIONALITY AND REGISTRATION MARKS			AIRCRAFT SERIAL NUMBER	4 CATEGORY
	N372BG	Gulfstream GV-SP (G550)		5164	Transport
5.	AUTHORITY AND BAS	IS FOR ISSUANCE			

This airworthiness certificate is issued pursuant to the Federal Aviation Act of 1958 and certifies that, as of the date of issuance, the aircraft to which issued has been inspected and found to conform to the type certificate therefor, to be in condition for safe operation, and has been shown to meet the requirements of the applicable comprehensive and detailed airworthiness code as provided by Annex 8 to the Convention on International Civil Aviation, except as noted herein Exceptions

Exemption 7946 25.813(e) Door Between Passenger Compartments Exemption 8004 25.901(c) Single Failure Criteria Single Failure Criteria Exemption 8142 25.901(c)

6. TERMS AND CONDITIONS

Unless sooner surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator, this airworthiness certificate is effective as long as the maintenance, preventative maintenance, and alterations are performed in accordance with Parts 21, 43, and 91 of the Federal Aviation Regulations, as appropriate, and the aircraft is registered in the United States.

DATE OF ISSUANCE

R 8/21/07

FAA REPRESENTATIVE

DESIGNATION NUMBER Emery P. Wiltse

DART-609001-GL

Any alteration, reproduction, or misuse of this certificate may be punishable by affine not exceeding \$1,000, or imprisonment not exceeding \$ years, or both. THIS CERTIFICATE MUST BE DISPLAYED IN THE AIRCRAFT IN ACCORDANCE WITH APPLICABLE FEDERAL AVIATION REGULATIONS

FAA Form 8100-2 (8-82)

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Berwind Corporation Kevin Wiig, Director of Aviation 5 Hog Island Road Philadelphia, PA 19153

9/17/2018

Mr. Mike Masterson FAA Flight Standards District Office Federal Aviation Administration 2 International Plaza, Suite 110 Philadelphia, PA 19113

Subject: Gulfstream G550, SN 5164 Registration Number Change

Dear Mr. Masterson:

Berwind Corporation is changing the Registration Number of our Gulfstream G550, serial number 5164. The previous registration number was N372BG. The new and current registration number is N272BG. We've completed FAA Form 8050-64 in support of this process. Additionally, the aircraft's logbook has the associated entry detailing the change along with the repaint of the registration number being now complete. We're requesting that a replacement Airworthiness Certificate be issued allowing us to continue the process and formally complete this process.

Should you have any questions please contact me at your convenience. I can be reached at my email address, kwiig@berwind.com, or on my cell phone at 267-408-3925. Thank you for your assistance.

Respectfully,

Kevin Wiig

Director of Aviation

BERWIND Aliftenne Logbook
N372BG - Gutfistream G550, SIN 5164

09 Aug 2018

Airframe Hours:

4710.2

Airframe Cyc/Lndgs:

2154

Effective August 9, 2018 this aircraft Registration changed from N372BG to N272BG. All work performed IAW Exterior Paint/Placard Layout Drawing for S/N 5164 Drawing# GC529060025 and Ident Strapping Drawing# CE513240411.

applicable manufacturer's maintenance manuals and current Federal Aviation performed. Administration Regulations and Approved for Return To Service with repect to work I certfy that all maintenance performed on this Aircraft has been in accordance with the

A. Andino

A&P #32221132

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ion, Extiadelphia, PA 19153, 215-365-5252

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Memorandum

Date:

19 September 2018

To:

AFS 750 Aircraft Registration

From:

Michael E. Masterson ASI, EA-17 Philadelphia FSDO

Subject:

Certificate of Airworthiness N272BG

Issued new Certificate of Airworthiness for Gulfstream GV-SP G-550 aircraft with registration # N272BG, S/N 5164, aircraft registration was changed from N372BG.

Attached:

New Certificate of Airworthiness for N272BG

Old Certificate of Airworthiness for N372BG

Signed statement of request.

Copy of logbook entry reflecting registration change on aircraft.

Michael E. Masterson ASI EA-17

610-595-1500 ext 254

Michael.masterson@faa.gov

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(3)
US Department
of Transportation
Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

OMB No. 2120-0020 Exp: 5/31/2018	Electronic Tracking Number
	For FAA Use Only

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		Name (As	showi	n on regi	stration certificate)				•		•	gistration ce	rtificate)	· · · · · · · · · · · · · · · · · · ·	
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Re		Alteration		Unit Ma			ke		Model					Serial No.		
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NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

3. Description of Work Accomplished		
(If more space is required, attach additional sheets.	Identify with aircraft nationality and registration mark and da	te work completed.)
	Gulfstream GV-SP(G550) sn5164	
	N372BG	12-20-2016
	Nationality and Registration Mark	Date

Installed Talon Aerospace LLC STC ST04177AT, issued September 24, 2014, in accordance with Master Document List (MDL) MDL-728-01 Revision 4 September 17, 2014 - as listed on the Approved Model List (AML) AML-728-01Revision 6 May 26, 2016 Item 38.

Installation Instructions Reference: LED Replacement General Purpose Lamps (INS-728-01 Revision 1 April 18, 2016) Part Numbers TAE0728-1 / TAE0728-1D, TAE 0628-1 / TAE0628-2, TAE0528-1 / TAE0528-1X / TAE0528-1W / TAE0528-2 / TAE0528-2W for Various Part 25 Aircraft

Instructions for Continued Airworthiness: Reference Talon Aerospace Maintenance Manual Supplement (AA-728-01) for Landing, Taxi, Turn, Scan and Logo Lights - Part Numbers TAE0728-1 / TAE0728-1D, TAE 0628-1 / TAE0628-2, TAE0528-1 / TAE0528-1X / TAE0528-1W / TAE0528-2 / TAE0528-2W for All Category Aircraft

Reference: Electrical Load Statement (ELA-728-01) Revision 3 Dated 14 April 2016 Power added as a result of this alteration is less than the power used by the OEM lamps. Therefore the power consumption as a result of this modification is negligible to the overall aircraft power system. No adjustments needed.

Reference: Structural Substantiation Report (STR-728-01) Revision 2 Dated 2 May 2016 There are no structural changes to the aircraft for installation of this modification.

Weight and Balance changes: are negligible Landing light + .235 lbs, Taxi light + .130 lbs. No W&B changes made at this time.

Reference: Component Maintenance Manual with Illustrated Parts List 33-42-02 Revision 3 Dated 29 April 2016 End.

Additional Sheets Are Attached

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Small Airplane Directorate Aircraft Certification Office 1701 Columbia Avenue Atlanta, Georgia 30337

Ken Martin Talon Aerospace, LLC 233 Cahaba Valley Parkway Pelham, AL 35124

Dear Mr. Martin:

We have completed our evaluation of your Supplemental Type Certificate, FAA Project Number ST14465AT-T, and find that you have satisfactorily demonstrated compliance with the applicable certification regulations. Accordingly, we have enclosed STC ST04177AT, which indicates our approval of the Installation of Talon Aerospace LED Replacement General Purpose Lamps.

This STC is official Federal Aviation Administration (FAA) approval of your modification and may be used to authorize identical modifications on other aircraft of the same model, subject to the limitations noted on the certificate. It may be transferred or otherwise made available to another party by means of a licensee arrangement in accordance with Federal Aviation Regulation (FAR) 21.47. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission. You are required to advise this office within 30 days after the transfer when you transfer or grant licensee rights to the STC in order that we may take the necessary recording or reissuance action.

As recipient of this approval, except as provided in FAR 21.3(d), you are required to report any failure, malfunction, or defect in any product or part manufactured by you that you have determined has resulted or could result in any of the occurrences listed in FAR 21.3(c). The report should be communicated initially by telephone to the Manager, Atlanta Aircraft Certification Office, telephone number (404) 474-5500, within 24 hours after it has been determined that the failure has occurred. In addition, written notification to the Certification Office, at the above address is required. FAA Form 8010-4 (Malfunction or Defect Report) or any other appropriate format is acceptable in transmitting the required details.

If you plan to manufacture replacement or modification parts for sale in conformance with approved data listed on the Certificate, you are required to comply with FAR 21.303. A Parts Manufacturer Approval (PMA) may be issued under the provisions of 14 FAR 21.303 when you submit a statement certifying you have established the Quality System as required by 14 FAR 21.307. Your statement may be in letter form, with reference to STC ST04177AT, and should be addressed to the Federal Aviation Administration, Small Airplane Directorate,

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Attention: Manager, Atlanta Manufacturing Inspection District Office, 1701 Columbia Avenue, Atlanta, Georgia 30337.

Please contact Bryan Long of this office for any questions or further discussion at (404) 474-5578.

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Melvin . Taylor, Manager,
Atlanta Aircraft Certification Office

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Muited States of America Bepartment of Transportation -- Mederal Abiation Administration

Supplemental Type Certificate

Number STOALTTAT

This certificate issued to

Talon Aerospace, LLC 233 Cahaba Valley Parkway Pelham, AL. 35124

exclifies that the change in the type design for the following product with the limitations and conditions therefor as succified heren meets the nieweethinss requirements of Part 25 of the Federal Aviation Begulatiens.

Criginal Second Type Sectificate Simber

See attached

Maker

FAA Approved Model List (AML)

Document Talon Aerospace AML-728-01 for a list of

Approved Airplane Models

Lescription of Type Design Change.

Installation of a Talon Aerospace LED Replacement General Purpose Lamps in accordance with Master Document List MDL-728-01, Revision 4, dated September 17, 2014, or later FAA approved revision.

Limitations and Conditions:

This approval should not be extended to other aircraft of this model on which other previously approved modifications are incorporated unless it is determined by the installer that the interrelationship between this change and any of those other previously approved modifications will produce no adverse affect upon the airworthiness of that airplane. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission. Instructions for Continued Airworthiness (ICA), Document Number AA-728-01, Revision 0, dated August 20, 2014, or later FAA accepted revision must be made available to the operator at the time of installation.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked or a termination date is otherwise established by the Administrator of the Federal Arciation Administration

Sute of application: October 18, 2013.

Sale reissued:

Date amended:

Sutressioner: September 24, 2014

Melvin D. Taylor, Manager,

Atlanta/Aircraft Certification Office

(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000. Or imprisonment not exceeding J years, or both.

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Letter of Authorization

This Letter of Authorization ("LOA") has been granted as of June 10th, 2016 to Berwind Aviation, having its principle place of business at 5 Hog Island Road, Philadelphia PA 19153 (herein referred to as "Customer") by Talon Aerospace LLC, having its principal place of business at 233 Cahaba Valley Parkway, Pelham, Alabama 35124 (herein after referred to as "Talon Aerospace") either or both of which may be referred to herein as a "Party" or the "Parties", respectively.

Whereas, Talon Aerospace is the holder of Supplemental Type Certificate (STC) ST04177AT.

Whereas, Customer desires to install and maintain LED sealed beam replacement lamps covered under the STC on customer aircraft.

Grant of Authorization & License to Use STC ST04177AT

In accordance with Title 14, Code of Federal Regulations (CFR) §21.120, Talon Aerospace authorizes Customer to use STC ST04177AT and its associated data package to alter aircraft as follows:

- Customer is authorized to install and maintain LED sealed beam replacement lamps including TAE0728-1, and TAE0528-1 as long as they are obtained from Talon's authorized distributor, Avio-Diepen, at standard list prices or special program pricing.
- Customer may install and maintain the lamps in accordance with this authorization at any of the Customer locations or at their assigned MROs.
- Authorization to use the STC is subject to the installation on aircraft listed on the latest revision of the Approved Model List included in the STC.
- Talon grants Customer a royalty free, worldwide, irrevocable license to use STC ST04177AT as long as used in accordance with this authorization.

In Witness Thereof, Talon Aerospace has executed this Agreement through their duly authorized representative.

Talon Aerospace LLC

By: Ken Martin

Ken Martin, Vice President Product Development

FAA APPROVED MODEL LIST (AML)

for INSTALLATION OF TALON AEROSPACE GENERAL PURPOSE LED LAMPS,

2W, TAE0428-1, TAE0428-1W, TAE0428-2, and TAE0428-2W, Bracket Assembly K428-B3-LGL-001 and K428-B3-LGL-002. MD88 Logo Light Kit K528-D9-LGL-001, K428-B5-LGL-001, K428-B5-LGL-001, K428-B5-LGL-001, K428-B5-LGL-001, K428-B5-LGL-001, K428-B7-LGL-002, K428-B5-LGL-003, K428-B5-LGL-004. K428-A3-LGL-001, K428-A3-LGL-001W, K428-B7-LGL-004. INCLUDING PART NUMBERS TAE0728-1, TAE0728-1D, TAE0628-1, TAE0628-2, TAE0528-1, TAE0528-1X, TAE0528-1W, TAE0528-2 and TAE0528-1 001, K428-B7-LGL-002, K428-B7-LGL-003, K438-B7-LGL-004

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	MAKE	
A300, Model B2-1A A300, Model B2-1C A300, Model B4-2C A300, Model B4-2C A300, Model B4-203 A300, Model B4-203 A300, Model B4-203 A310, Model B221 A310, Model 221 A310, Model 222 A310, Model 324 A310, Model 322 A310, Model 325 A310, Model B4-601 A300, Model B4-605R A300, Model B4-622R A300, Model F4-622R A300, Model F4-605R Variant F A300, Model B4-622	MODEL	AIDCDAFF
	NUMBER	ORIGINAL TYPE
Amend. 25-1 through 25-123	ALTERATION	CERTIFICATION
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Airbus	AIRCRAFT MAKE	
A318 Series A318-111, A318-121, A318-121, A318-122, A319-111, A319-113, A319-113, A319-113, A319-133, A319-131, A319-131, A320-211, A320-211, A320-211, A320-214 A320-233 A320-214 A320-233 A320-233 A320-231 A321-211, A321-111, A321-111, A321-121, A321-213, A321-211, A321-213, A321-213, A321-213, A321-213, A321-213, A321-231	AIRCRAFT MODEL	
A28NM	CERTIFICATE NUMBER	ORIGINAL TYPE
14 CFR Part 25, Amend. 25-1 through 25-123	BASIS FOR ALTERATION	CERTIFICATION
INS-728-01	DRAWING	INSTAL
Rev 0, 3/14/14	REVISION AND DATE	INSTALLATION INSTRUCTIONS
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	AMENDMENT DATE	<u> </u>

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Airbus	Airbus	AIRCRAFT MAKE
A340-200 Series A340-211, A340-212, A340-213 A340-300 Series A340-311, A340-312, A340-313 A340-500 Series A340-500 Series A340-600 series	A330-200 Series A330-201, A330-203, A330-223, A330-2243 A330-2200 Freighter Series A330-223F, A330-243F A330-300 Series A330-301, A330-302, A330-303, A330-321, A330-322, A330-323, A330-323, A330-323, A330-341, A330-341, A330-343	AIRCRAFT MODEL
A43NM	A46NM	ORIGINAL TYPE CERTIFICATE NUMBER
14 CFR Part 25, Amend, 25-1 through 25-123	14 CFR Part 25, Amend. 25-1 through 25-123	CERTIFICATION BASIS FOR ALTERATION
INS-728-01 A3-004	INS-728-01 A3-004	INSTAI INSTRI DRAWING NUMBER
Rev 0, 3/14/14 Rev 1, 4/18/16	Rev 0, 3/14/14 Rev 1, 4/18/16	INSTALLATION INSTRUCTIONS REVISION ER AND DATE
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ATR72-212A	ATR72-212	ATR72-211	ATR72-202	ATR72-102	ATR72-201	ATR72-101	ATR42-500	ATR42-320	ATR42-300	ATR42-200		MODEL	AIRCRAFT		
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								through 25-123	Amend. 25-1	14 CFR Part 25,		ALTERATION	BASIS FOR	CERTIFICATION	
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Beecherali	MAKE	AIRCRAFT
HS. 125 Series F3B HS. 125 Series F3B/RA BH. 125 Series 400A DH. 125 Series 400A HS. 125 Series 400B/I HS. 125 Series 400B/I HS. 125 Series 401B HS. 125 Series 403B HS. 125 Series 403B HS. 125 Series 600B HS. 125 Series F400B HS. 125 Series F400B/I	MODEL	AIRCRAFT
ASEC	TYPE CERTIFICATE NUMBER	ORIGINAL
14 CFR Part 25, Amend. 25-1 through 25-123	BASIS FOR ALTERATION	CERTIFICATION
NS-/28-01	DRAWING NUMBER	INSTAL
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Beechcraft	MAKE	AIRCRAFT
HS.125 Series F600B HS.125 Series 700A HS.125 Series 700B BAe.125 Series 800A BAe.125 Series 800A (C-29A) BAe.125 Series 800A (U-125) BAe.125 Series 800B BAe.125 Series 1000A F3B BAe.125 Series 1000A F3B BAe.125 Series 1000A F3B BAe.125 Hawker 800 Hawker 800 Hawker 800 Hawker 800XP Hawker 800XP Hawker 800XP Hawker 900XP	MODEL	AIRCRAFT
A3EU	TYPE CERTIFICATE NUMBER	ORIGINAL
14 CFR Part 25, Amend. 25-1 through 25-123	BASIS FOR ALTERATION	CERTIFICATION
INS-728-01	DRAWING NUMBER	INSTALLATION INSTRUCTIONS
Rev 0, 3/14/14	REVISION AND DATE	
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	AMENDMENT	A.

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9 8 7 TEM	AIRCRAFT MAKE Boeing Boeing	AIRCRAFT MODEL 707-100 Long Body 707-100B Long Body 707-100B Short Body 707-300 Series 707-300 Series 707-300 Series 707-300 Series	ORIGINAL TYPE CERTIFICATE NUMBER 4A21	CERTIFICATION BASIS FOR ALTERATION 14 CFR Part 25, Amend. 25-1 through 25-123 14 CFR Part 25, Amend. 25-1 through 25-123		INSTAI INSTRI DRAWING NUMBER INS-728-01 INS-728-01	INSTALLATION INSTRUCTIONS PRAWING REVISION AND DATE Rev 0, 3/14/14 Rev 0, 3/14/14 Rev 0, 3/14/14 Rev 0, 3/14/14
Boci	3.0	707-300 Series 707-400 Series 707-300B Series 707-300C Series	4/26	14 CFR Part 25, Amend. 25-1 through 25-123	INS-728-01	Rev 0. 3/1	4/14
	Boeing	737-100 Series 737-200 Series 737-200C Series 737-300 Series 737-400 Series 737-500 Series 737-700 Series 737-700 Series 737-700 Series 737-900 Series 737-900 Series 737-900 Series	A16WE	14 CFR Part 25, Amend. 25-1 through 25-123	INS-728-01 B3-004	Rev 0, 3/14/14 Rev 1, 8/11/15	/14/14

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Boeing	Boeing	Boeing	AIRCRAFT MAKE
767-200 Series, 767-300 Series, 767-300F Series, 767-400ER Series	757-200Series, 757-200PF Series, 757-200CB Series, 757-300 Series	747-100 Series 747-200B Series 747-200F Series 747-200C Series 747-200C Series 747SP Series 747-100B Series 747-100B Sud Series 747-400B Sud Series 747-400D Series 747-400F Series 747-400F Series 747-400F Series	AIRCRAFT MODEL
AINM	A2NM	A20WE	ORIGINAL TYPE CERTIFICATE NUMBER
14 CFR Part 25. Amend. 25-1 through 25-123	14 CFR Part 25, Amend. 25-1 through 25-123	14 CFR Pan 25, Amend. 25-1 through 25-123	CERTIFICATION BASIS FOR ALTERATION
INS-728-01	INS-728-01 B5-004	INS-728-01	INSTALI INSTRU DRAWING NUMBER
Rev 0, 3/14/14	Rev 0, 3/14/14 Rev 1, 9/10/15	Rev 0, 3/14/14	INSTALLATION INSTRUCTIONS ING REVISION ER AND DATE
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Boeing			Boeing	MAKE	AIRCRAFT
787-8	Series 777-200LR Series 777F Series	777-300 Series 777-300ER	777-200 Series	MODEL	AIRCRAFT
T00021SE			T00001SE	TYPE CERTIFICATE NUMBER	ORIGINAL
14 CFR Part 25, Amend. 25-1 through 25-123		Amend, 25-1 through 25-123	14 CFR Part 25,	BASIS FOR ALTERATION	CERTIFICATION
INS-728-01		B7-004	INS-728-01	DRAWING NUMBER	INSTALLATION INSTRUCTIONS
Rev 0, 3/14/14		Rev 1, 4/18/16	Rev 0, 3/14/14	REVISION AND DATE	LATION CTIONS
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	Boeing	MAKE	AIRCRAFT
DC-9-13 DC-9-14 DC-9-15F DC-9-15F DC-9-31 DC-9-31 DC-9-32 (VC9C) DC-9-32F (VC9C) DC-9-33F DC-9-33F DC-9-34 DC-9-31 (MD-81) DC-9-81 (MD-82) DC-9-82 (MD-82) DC-9-83 (MD-83) DC-9-87 (MD-83) MD-88 MD-90-30	DC-9-11 DC-9-12	MODEL	AIRCRAFT
	A6WE	TYPE CERTIFICATE NUMBER	ORIGINAL.
through 25-123	14 CFR Part 25, Amend, 25-1	BASIS FOR ALTERATION	CERTIFICATION
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Rev 0, 8/20/15	Rev 0, 3/14/14	REVISION AND DATE	CTIONS
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		AMENDMENT DATE	AML



	16		TEM
	Boeing	MAKE	AIRCRAFT
DC-10-15 DC-10-30 DC-10-30F DC-10-40 DC-10-40F MD-10-10F MD-10-30F MD-11 MD-11F	DC-10-10F	MODEL	AIRCRAFT
	A22WE	TYPE CERTIFICATE NUMBER	ORIGINAL
through 25-123	14 CFR Part 25, Amend, 25-1	BASIS FOR ALTERATION	CERTIFICATION
	INS-728-01	DRAWING NUMBER	INSTAL INSTRU
	Rev 0, 3/14/14	REVISION AND DATE	INSTALLATION INSTRUCTIONS
	N/N	SUPPLEMENT NUMBER/DATE	AFM
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Bombardier	AIRCRAFT MAKE	
CL-600-1 A11 (CL-600) CL-600-2A12 (CL-601) CL-600-2B16 (CL-601-3A Variant) CL-600-2B16 (CL-601-3R Variant) CL-600-2B16 (CL-600-2B16 (CL-600-2B19 (Regional Jet Series 700, 701 & 702) CL-600-2D15 (Regional Jet Series 705) CL-600-2D24 (Regional Jet Series 900) CL-600-2E25 (Regional Jet Series 900)	AIRCRAFT MODEL	
\21E\	CERTIFICATE NUMBER	ORIGINAL TYPE
14 CFR Part 25, Amend. 25-1 through 25-123	BASIS FOR ALTERATION	CERTIFICATION
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Bombardier	Bombardier	Bombardier	Bombardier	AIRCRAFT MAKE	
CL-215-1A10 CL-215-6B11 (CL-215T Variant) CL215-6B11 (CL-415 Variant)	DHC-8-100 Scries DHC-8-200 Scries DHC-8-300 Series DHC-8-400 Scries	BD-700-1A10 BD-700-1A11	BD-100-1A10	AIRCRAFT MODEL	
A14EA	WNEIV	T00003NY	T.0000Y	ORIGINAL TYPE CERTIFICATE NUMBER	
14 CFR Part 25, Amend. 25-1 through 25-123	14 CFR Part 25, Amend. 25-1 through 25-123	14 CFR Part 25, Amend. 25-1 through 25-123	14 CFR Part 25, Amend. 25-1 through 25-123	CERTIFICATION BASIS FOR ALTERATION	
INS-728-01	INS-728-01	INS-728-01	INS-728-01	DRAWING NUMBER	INSTALLATION INSTRUCTIONS
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			through 25-123		Mystere-		
			Amend. 25-1		Falcon 50	Aviation	
4	Rev 0, 3/14/14	INS-728-01	14 CFR Part 25,	A46EU	Mystere-	Dassault	29
			through 25-123			•	
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4	Rev 0 3/14/14	INS-728-01	14 CFR Part 25	A SOUM	Falcon 7X	Dassault	3,5
			hrough 25-123		Janaer	Aviation	
14	Rcv 0, 3/14/14	INS-728-01	14 CFR Part 25.	A50NM	Falcon 2000	Dassault	27
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			Amend, 25-1			Aviation	
11/	Rev 0, 3/14/14	INS-728-01	14 CFR Part 25,	A33EU	Falcon 10	Dassault	26
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14 ÇFR Part 25; Amend, 25-1 through 25-123	14 CFR Part 25, Amend. 25-1 through 25-123	CERTIFICATION BASIS FOR ALTERATION
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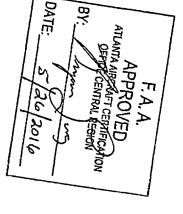
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Fellowship F.28 Mark 1000 F.28 Mark 2000 F.28 Mark 3000 F.28 Mark 4000 F.28 Mark 0100 F.28 Mark 0100	F.27 Mark 100 F.27 Mark 200 F.27 Mark 300 F.27 Mark 400 F.27 Mark 500 F.27 Mark 600 F.27 Mark 700 F.27 Mark 700	ERJ 190-100 STD ERJ 190-100 LR ERJ 190-100 ERJ 190-200 ERJ 190-200 STD ERJ 190-200 LR ERJ 190-200 IGW	AIRCRAFT MODEL
A-817	A-817	AS7NM	ORIGINAL TYPE CERTIFICATE NUMBER
14 CFR Part 25, Amend. 25-1 through 25-123	14 CFR Part 25, Amend. 25-1 through 25-123	14 CFR Part 25, Amend. 25-1 through 25-123	CERTIFICATION BASIS FOR ALTERATION
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	N/A	Rev 0, 3/14/14	INS-728-01	14 CFR Part 25, Amend. 25-1 through 25-123	A12EA	G-1159A G-1159B G-IV GV-SP GV-X	Gulfstream	رب چ
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14 CFR Part 25, Amend. 25-1 through 25-123	14 CFR Part 25, Amend. 25-1 through 25-123	BASIS FOR ALTERATION	CERTIFICATION
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Saab AB, Saab Aerosystems	Saab AB, Saab Acrosystems	AIRCRAFT MAKE	
SAAB 2000	340A (SAAB/SF340 A) SAAB 340B	AIRCRAFT MODEL	
A47NM	AS2EU	ORIGINAL TYPE CERTIFICATE NUMBER	
14 CFR Part 25, Amend. 25-1 through 25-123	14 CFR Part 25, Amend. 25-1 through 25-123	CERTIFICATION BASIS FOR ALTERATION	
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TALON AEROSPACE Installation Instructions LED Replacement General Purpose Lamps

TAE0728-1, TAE0728-1D
TAE0628-1 / TAE0628-2
TAE0528-1 / TAE0528-1X / TAE0528-2 / TAE0528-1W / TAE0528-2W

For Various Part 25 Aircraft

Prepared By: <u>Few Marks</u>	_ Date: <u>4/18/14</u> _
Checked by: Tyh Hohm	Date: 4/18/16
Approved by:	Date: <u>4118/16</u>

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RECORD OF REVISIONS

Rev	Date	Pages Affected	Description Of Revision	Approved By
0	3/14/14	All	Initial Release	Prepared by: Ken Martin Checked by: Tyler Henderson Approved by: Jose Perez
1	See Front Page	3	Added Part Numbers TAE0728-1D and TAE0528-1X.	Prepared by: Ken Martin Checked by: Tyler Henderson . Approved by: Jose Perez

INSTALLATION INSTRUCTIONS

INTRODUCTION

The Talon Aerospace LED replacement lamps are "plug and play" assemblies that are one for one replacements for the existing PAR 36, PAR 46 and PAR 64 incandescent lamps currently in use as landing, turn / taxi, engine and wing scan, and logo lights. They are essentially replacement lamps that are repairable and designed to last the life of the aircraft.

Refer to the particular Aircraft Maintenance Manual for installation in the various positions used for installation and aiming of the lamps (if required).

When applying torque to the screws connecting the wire to the terminals, hold the terminal with an open end wrench to avoid breaking the seal.

Following is a list of the part numbers and applications;

TAE0728-1	PAR 64. Landing light for large air transport aircraft. Spot beam
TAE0728-1D	PAR 64. Taxi/Take-Off light for large transport aircraft. Dual mode, High and Low Beam.
TAE0628-1	PAR 46. Landing light for some commuter and business jets. Spot beam.
TAE0628-2	PAR 46. Taxi/Turn light. Wide horizontal beam.
TAE0528-1	PAR 36. Cool white. Spot beam.
TAE0528-1X	PAR 36. Cool white. Spot Beam. High intensity.
TAE0528-2	PAR36. Cool white. Wide horizontal beam.
TAE0528-1W	PAR 36. Warm white. Spot beam.
TAE0528-2W	PAR 36. Warm white. Wide horizontal beam.

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INSTRUCTIONS FOR CONTINUED AIRWORTHINESS

Maintenance Manual Supplement

For

Talon Aerospace

Landing, Taxi, Turn, Scan and Logo Lights

Part Numbers

TAE0728-1/ TAE0728-1D

TAE0628-1 / TAE0628-2

TAE0528-1 / TAE0528-1X / TAE0528-1W / TAE0528-2 / TAE0528-2W

For

All Transport Category Aircraft

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INSTRUCTIONS FOR CONTINUED AIRWORTHINESS

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Rev	Date	Pages Affected	Description Of Revision	Approved By
0	8/20/14	All	Initial Release	Prepared by: Ken Martin Checked by: Tyler Henderson Approved by: Jose Perez
1	12/19/14	4	Paragraph 6. Inserted statement on inspection intervals.	Prepared by: Ken Martin Checked by: Tyler Henderson Approved by: Jose Perez
2	4/7/15	5 6	Note 1 revised airworthiness limitation statement Paragraph 4 and 5 revised statement on inspection intervals.	Prepared by: Ken Martin Checked by: Tyler Henderson Approved by: Jose Perez
3	1/15/16	1 6 8	Added TAE0728-1D and TAE0528-1X. Paragraph 4. Inspection Intervals, changed to C Check or 24 months. Revised wording, Paragraph 2.	Prepared by: Ken Martin Checked by: Tyler Henderson Approved by: Jose Perez

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INTRODUCTION

This manual contains Instructions for Continued Airworthiness for the LED Replacement PAR 36, PAR 46 and PAR 64 Lamps, used as Landing, Taxi, Turn, Scan and Logo Lights on various air transport aircraft. These lamps include the following part numbers; TAE0728-1 (landing), TAE0628-1 (landing), TAE0628-2, (taxi) and TAE0528-1 and -1W, as well as TAE0528-2 and -2W (turn, taxi, engine/wing scan and logo) as manufactured by TALON AEROSPACE.

These Instructions are provided as required by the Code of Federal Regulations Title 14 Part 25 Section 25.1529 and Part 21 Section 21.50(b).

The instructions apply only to the component part numbers listed above. Be sure the part numbers on the LED lamps are listed on the title page before performing any of these instructions.

In addition to supplying all owners of the Landing, Taxi, Turn, Scan and Logo Lights with this initial Maintenance Manual Supplement, Talon Aerospace will provide all revisions and updates to this manual to the owner in a timely manner as mandated by FAA Order 8110.54A.

Initial Maintenance Manual Supplements and all subsequent revisions and updates will be mailed in hard copy form to the Landing, Taxi, Turn, Scan and Logo Lights owner, and/or provided in electronic format and directed to departments or personnel as specified by the owner.

If you feel that this manual does not meet your needs per the above mentioned Regulations, feel free to contact Talon Aerospace for clarification or additional information.

TALON AEROSPACE (4FSN4) Product Support 233 Cahaba Valley Parkway Pelham, Alabama 35124 Telephone: (205) 403-6589 FAX: (205) 403-6590

support@talonaerospace.com

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AIRWORTHINESS LIMITATIONS

1. Notes.

- A. The LED Landing, Taxi, Turn, Engine-Wing Scan and Logo Lights should be removed and the LED Board assemblies replaced if light intensity is no longer adequate for the intended purpose as reported by flight crews or maintenance personnel.
- B. "The Airworthiness Limitations section is FAA-approved and specifies inspections and other maintenance required under §43.16 and § 91.403 of the Federal Aviation Regulations, unless an alternative program has been FAA approved".

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DESCRIPTION AND OPERATION

- 1. **Description.** The LED Landing, Taxi, Turn, Engine-Wing Scan and Logo Lights consist of the following components; aluminum housing, a power supply to convert the AC voltage to DC voltage, the LED assembly, the optical assembly, and a lens and bezel, along with a back cap, a harness, gaskets and hardware to seal the units against the environment. These units connect to the aircraft electrical system as the original incandescent lamps do. They are "plug and play", requiring no alteration to the aircraft. In addition to providing longer life than the incandescent lamps that they replace, these units are repairable.
- 2. Basic Control and Operation. The Par 64 Landing and Par 46 Landing/Taxi Lights operate on either 26 VAC or 28 VDC, while the Par 36 Runway Turn Off/Engine/Wing Scan/Logo Light operates on 13 or 26 VAC or 28VDC. All lamps are controlled by the same switches in the cockpit that control the current lamps. There are no special procedures or limitations that apply to the operation of the Landing, Taxi, Turn, Scan and Logo Lamps.
- 3. Warnings and Cautions. There no special warnings and cautions concerning the Talon Aerospace Landing, Taxi, Turn, Scan and Logo Lights, These lights use LED technology and eliminate the high temperatures associated with incandescent lamps, thereby eliminating any injuries to personnel associated with the incandescent lamps.
- 4. Inspections. Inspection intervals: Inspections to be performed during Aircraft C Check interval or at 24 months of operation.

With the units installed in their respective housings, inspect the Talon Aerospace LED Landing, Taxi, Turn, Scan and Logo Lights for the following:

Lenses cracked, or if nicked or scratched deeper than .03 inches Any corrosion is found deeper than .03 inches, or covers more than 20 percent of the metal surface 3 or more individual LEDs failed on the LED board If the part fails or if any damage is noted during inspections requiring removal

If any of the above conditions are found, remove the Talon LED Landing, Taxi, Turn, Scan and Logo Lights and route to repair shop for repair in accordance with Talon Aerospace Component Maintenance Manual & IPL 33-42-02.

5. Hard Landing, Hail Damage or Bird Strike. Perform an inspection for damage as in paragraph 4 and action accordingly.

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6.	Leading Particulars.	
	DESCRIPTION	DATA
	Landing Light TAE0728-1 / TAE0728-1D	
	Dimensions:	
	Diameter	8.0 In. (203.2 mm)
	Height	2.75 In. (69.85 mm)
	Weight (Approx.):	2.00 lb. (0.907 kg)
	Voltage:	28 VDC or 26 VAC
	Landing / Taxi Light TAE0628-1 and TAE0628	H-2
	Dimensions:	
	Diameter	5.75 In. (146.05 mm)
	Height	2.75 In. (69.85 mm)
	Weight (Approx.):	0.99 lb. (0.449 kg)
	Voltage:	28 VDC or 26 VAC
TΑ	Taxi / Turn / Engine-Wing Scan / Logo Light T LE0528-2W	TAE0528-1, TAE0528-1X, TAE0528-1W, TAE0528-2 and
	Dimensions:	
	Diameter	4.5 ln. (114.3 mm)
	Height	2.1 ln. (53.34 mm)
	-	· · · ·
	Weight (Approx.):	0.60 lb. (0.27 kg)
	Voltage:	28 VDC or 13/26 VAC

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REMOVAL AND REPLACEMENT

Removal and replacement of the Talon Aerospace LED Landing, Taxi, Turn, Scan and Log Lights is to be accomplished in accordance with the applicable Aircraft Maintenance Manual.

TESTING AND FAULT ISOLATION

After removal from the aircraft due to failure, Testing and Fault Isolation of the Talon Aerospace LED Landing, Taxi, Turn, Scan and Logo Lights is to be accomplished in accordance with the Talon Aerospace Component Maintenance Manual 33-42-02.

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Component Maintenance Manual With Illustrated Parts List

For

Talon Aerospace

LED Landing, Taxi/Take-Off, Taxi, Turn, Scan and Logo Lights

Part Numbers

TAE0728-1 / TAE0728-1D

TAE0628-1 / TAE0628-2

TAE0528-1 / TAE0528-1W / TAE0528-1X / TAE0528-2 / TAE0528-2W

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33-42-02

29 April 2016 Revision 3

RECORD OF REVISIONS

Retain this record in the front of the manual. On receipt of revisions, insert revised pages in manual. Enter revision number, date issued, date inserted and initials.

REVISION NUMBER	DATE ISSUED	DATE INSERTED	ВУ
0	7/27/2014		
1	4/8/2015		
2	8/13/2015		
3	4/29/2016		
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RECORD OF TEMPORARY REVISIONS

REVISION NUMBER	REVISION DATE	DATE INSERTED	ВУ	DATE REMOVED	ВУ

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SERVICE BULLETINS

SERVICE BULLETIN NUMBER	REVISION NUMBER	DATE ADDED TO MANUAL
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Service Bulletins Page SBL-1

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Disassembly	301
Cleaning	401
Repair	601
Assembly (Including Storage)	701
Special Tools, Fixtures and Equipment	801
Illustrated Parts List	901

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INTRODUCTION

This manual contains Instruction for Continued Airworthiness with Illustrated Parts List (IPL) for the LED Replacement PAR 36, PAR 46 and PAR 64 Lamps, used as Landing, Taxi, Turn, Scan and Logo Lights on various air transport aircraft. These LED lamps include the following part numbers; TAE0728-1 (landing), TAE0728-1D (dual mode taxi/lake-off), TAE0628-1 (landing), TAE0528-1X (landing), TAE0628-2, (taxi) and TAE0528-1 and -1W as well as TAE0528-2 and -2W (turn, taxi, engine/wing scan and logo) as manufactured by TALON AEROSPACE.

These Instructions are provided as required by the Code of Federal Regulations Title 14 Part 25 Section 25.1529 and Appendix H to Part 25.

The instructions apply only to the component part numbers listed above. Be sure the part numbers on the LED lamps are listed on the title page before performing any of these instructions.

In addition to supplying all owners of the LED Landing, Taxi, Turn, Scan and Logo Lights with this initial Component Maintenance Manual, Talon Aerospace will provide all revisions and updates to this manual to the owner in a timely manner as mandated by FAA Order 8110.54.

Initial Component Maintenance Manuals and all subsequent revisions and updates will be mailed in hard copy form to the LED Landing, Taxi, Turn, Scan and Logo Lights owner, and/or provided in electronic format and directed to departments or personnel as specified by the owner.

If you feel that this manual does not meet your needs per the above mentioned Regulations, feel free to contact Talon Aerospace for clarification or additional information.

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support@talonaerospace.com



AIRWORTHINESS LIMITATIONS

1. Notes.

- A. The LED Landing, Taxi, Turn, Engine-Wing Scan and Logo Lights should be removed and the LED Board assemblies replaced if light intensity is no longer adequate for the intended purpose as reported by flight crews or maintenance personnel.
- B. "The Airworthiness Limitations section is FAA-approved and specifies inspections and other maintenance required under §43.16 and § 91.403 of the Federal Aviation Regulations, unless an alternative program has been FAA approved".

DESCRIPTION AND OPERATION

- **1. Description.** (Refer to Figure 1, 2 and 3). The LED Landing, Taxi, Turn, Engine-Wing Scan and Logo Lights consist of the following components; aluminum housing, a power supply to convert the AC voltage to DC voltage, the LED assembly, the optical assembly, and a lens and bezel, along with a back cap, a harness, gaskets and hardware to seal the units against the environment. These units connect to the aircraft electrical system as the original incandescent lamps do. They are "plug and play", requiring no alteration to the aircraft.
- **2. Basic Control and Operation.** The Par 64 LED Landing and Par 46 LED Landing/ Taxi Lights operate on either 26 VAC or 28 VDC, while the Par 36 LED Runway Turn Off/Engine/Wing Scan/Logo Light operates on 13 or 26 VAC or 28 VDC. All lamps are controlled by the same switches in the cockpit that control the current lamps. There are no special procedures or limitations that apply to the operation of the LED Landing, Taxi, Turn, Scan and Logo.

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DESCRIPTION AND OPERATION

3. Leading Particulars.

<u>DESCRIPTION</u> <u>DATA</u>

LED Landing Light TAE0728-1 and TAE0728-1D

Dimensions:

Diameter 8.0 In. (203.2 mm)
Height 2.75 In. (69.85 mm)

Weight (Approx): 2.00 lb (0.907 kg)

Voltage: 28 VDC or 26 VAC

LED Landing / Taxi Light TAE0628-1 and TAE0628-2

Dimensions:

Diameter 5.75 In. (146.05 mm)

Height 2.75 In. (69.85 mm)

Weight (Approx): 0.99 lb (0.449 kg)

Voltage: 28 VDC or 26 VAC

LED Landing/Taxi / Turn / Engine-Wing Scan / Logo Light TAE0528-1, TAE0528-1W, TAE0528-1X, TAE0528-2 and TAE0528-2W

Dimensions:

Diameter 4.5 In. (114.3 mm)

Height 2.1 In. (53.34 mm)

Weight (Approx): 0.60 lb (0.27 kg)

Voltage: 28 VDC or 13/26 VAC

TALON AEROSPACE COMPONENT MAINTENANCE MANUAL LANDING / TAXI/TAKE-OFF/TAXI / TURN / SCAN / LOGO LIGHTS

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TESTING AND FAULT ISOLATION

1. Notes.

- A. This section outlines the proper testing procedures for the LED Landing, Taxi, Turn, Scan and Logo Lamps.
- B. Tests should be conducted in the order below and all failed items should be replaced or repaired before continuing testing.
- C. Refer to Table 901 on page 801 for a list of tools, fixtures and equipment needed for testing. Equivalent substitutes are acceptable.

2. Pretest Procedures.

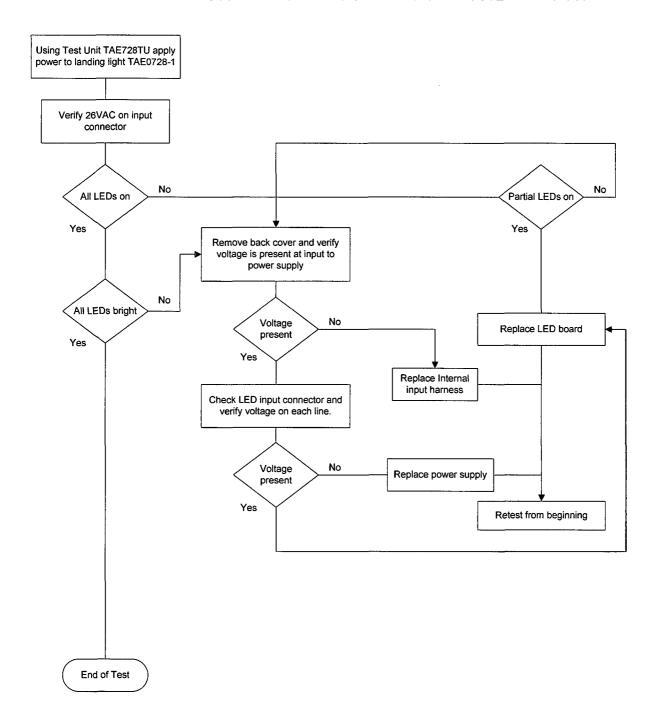
- A. Review any shipping tags or forms to determine cause for removal from aircraft.
- B. Wipe the LED Landing, Taxi, Turn, Scan and Logo Lamps with a clean cloth to remove dirt, dust and foreign material.
- C. Visually inspect the component for signs of missing, broken, bent or cracked parts. Damaged parts shall be replaced before the component light is tested.

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TESTING AND FAULT ISOLATION

3. Testing of LED Landing Light TAE0728-1.

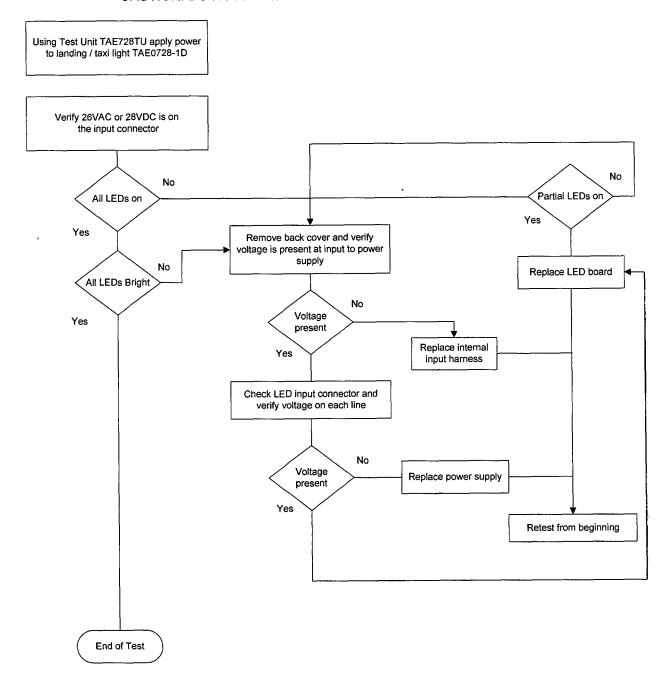
NOTE: Refer to IPL Figure 1 for an illustration. Numbers in parentheses () refer to item numbers on the illustration. CAUTION: DO NOT APPLY POWER TO UNIT UNTIL INSTRUCTED TO DO SO.



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4. Testing of LED Dual Mode Taxi/Take-Off Light TAE0728-1D.

NOTE: Refer to IPL Figure 1 for an illustration. Numbers in parentheses () refer to item numbers on the illustration. CAUTION: DO NOT APPLY POWER TO UNIT UNTIL INSTRUCTED TO DO SO.

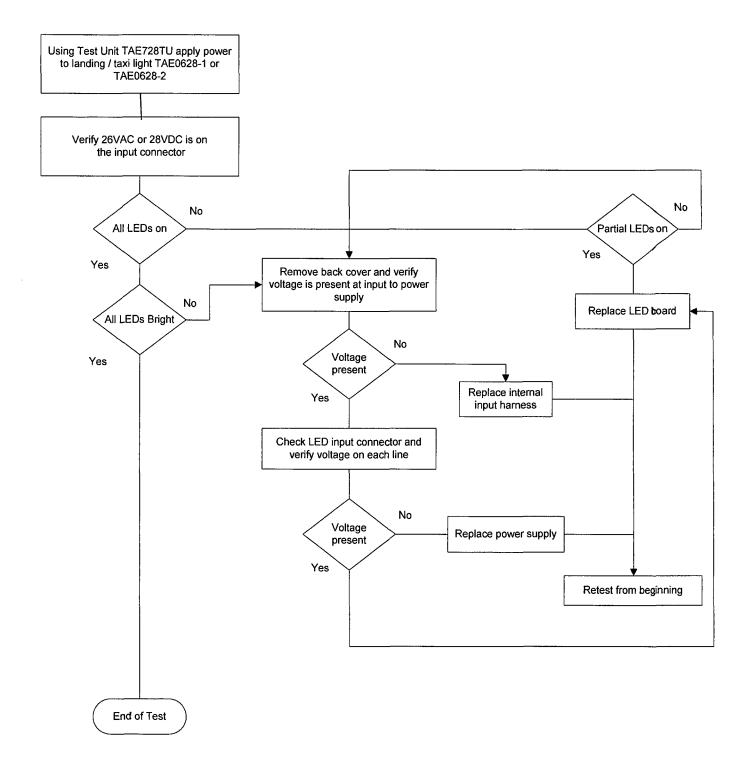


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5. Testing of LED Landing / Taxi Light TAE0628-1 or TAE0628-2.

NOTE: Refer to IPL Figure 2 for an illustration. Numbers in parentheses () refer to item numbers on the illustration.

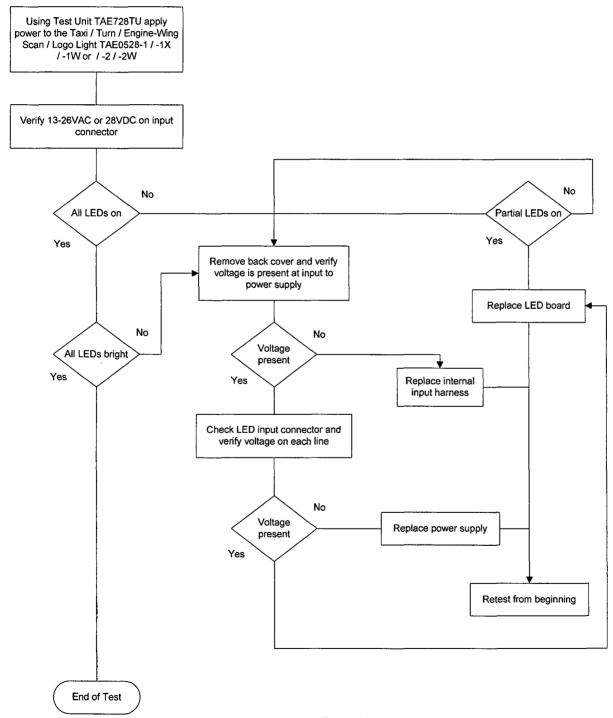
CAUTION: DO NOT APPLY POWER TO UNIT UNTIL INSTRUCTED TO DO SO.



6. <u>Testing of LED Landing / Taxi / Turn / Engine-Wing Scan/ Logo Light TAE0528-1, TAE0528-1X, TAE0528-1W, TAE0528-2, TAE0528-2W</u>

NOTE: Refer to IPL Figure 3 for an illustration. Numbers in parentheses () refer to item numbers on the illustration.

CAUTION: DO NOT APPLY POWER TO UNIT UNTIL INSTRUCTED TO DO SO.



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DISASSEMBLY- LED Landing Light TAE0728-1 and Taxi/Take-Off Light TAE0728-1D

- A. This section outlines the proper method of disassembly for the LED Landing Light and Taxi/Take-Off Light.
- B. Disassemble only to the extent required to replace defective parts.
- C. Refer to the Table on page 903 for a list of materials needed for assembly (including storage). Equivalent substitutes are acceptable.

2. Disassembly.

CAUTION: OBSERVE STANDARD ESD PROCEDURES WHILE HANDLING THE LED LANDING AND TAXI/TAKE-OFF LIGHT IN A DISASSEMBLED STATE.

NOTE: Refer to IPL Figure 1 for an illustration. Numbers in parentheses () refer to item numbers on the illustration.

- A. Before disassembly, wipe exterior surfaces with a clean cloth to remove dirt, dust and other foreign materials.
- B. When disassembling LED Landing and Taxi/Take-Off Light (refer to IPL Fig.1 page 902 and 903 for more information) unscrew Bezel (2) from housing (5) using TAE728-100 Spanning Wrench. This will allow the Lens (1), Optics Assembly (3) and Gasket (11) to be removed.
- C. To remove the Power Supply Assy. (6) remove Screws (16) from Cap (7), then Screws (15). This allows the Power Supply Assy. (6), Heat Thermal, PWS (10), Cap (7), and Back Gasket (12) to be removed.
- D. To remove LED Board (4) remove Screws (15) and pull up the LED Board (4) and Heat Thermal (9).
- E. To remove Harness (13) from Brass Standoff (18) remove Locknuts (19). Pull Shoulder Washer (21) and flat washer (20) from Standoff (18)..
- F. To remove Key (8) remove Screw (16)



DISASSEMBLY- LED Landing / Taxi Light TAE0628-1 and TAE0628-2

- A. This section outlines the proper method of disassembly for the LED Landing / Taxi Light.
- B. Disassemble only to the extent required to replace defective parts.
- C. Refer to the Table on page 905 for a list of materials needed for assembly (including storage). Equivalent substitutes are acceptable.

2. Disassembly.

CAUTION: OBSERVE STANDARD ESD PROCEDURES WHILE HANDLING THE LED LANDING / TAXI LIGHT ASSEMBLY IN A DISASSEMBLED STATE.

NOTE: Refer to IPL Figure 2 for an illustration. Numbers in parentheses () refer to item numbers on the illustration.

- Before disassembly, wipe exterior surfaces with a clean cloth to remove dirt, dust and other foreign materials.
- B. When disassembling LED Landing/Taxi Light (refer to IPL Fig.2 page 904 and 905 for more information) unscrew Bezel (2) from housing (5) using TAE628-100 Spanning Wrench. This will allow the Lens (1), Optics Assembly (3) and Gasket (10) to be removed.
- C. To remove the Power Supply Assy (6) remove Screws (16) to remove Cap (7), then Screws (15). This allows the Power Supply Assy. (6), Heat Thermal, PWS (12), Cap (7), and Back Gasket (11) to be removed.
- D. To remove LED Board (4) remove Screws (15) and pull up LED Board (4) and Heat Thermal (9).
- E. To remove Harness (13) from Brass Standoff (21) remove Locknuts (18). Pull Shoulder Washer (19) and flat washer (20) from Standoff (21).
- F. To remove Key (8) remove Screw (17)

DISASSEMBLY- LED Landing / Taxi / Turn / Engine-Wing Scan / Logo Light TAE 0528-1, TAE0528-1X, TAE0528-1W, TAE0528-2 and TAE0528-2W

- A. This section outlines the proper method of disassembly for the LED Landing / Taxi / Turn / Engine-Wing Scan / Logo Light.
- B. Disassemble only to the extent required to replace defective parts.
- C. Refer to the Table on page 907 for a list of materials needed for assembly (including storage). Equivalent substitutes are acceptable.

2. Disassembly.

CAUTION: OBSERVE STANDARD ESD PROCEDURES WHILE HANDLING THE LED LANDING / TAXI / TURN / SCAN / LOGO LIGHT IN A DISASSEMBLED STATE.

NOTE: Refer to IPL Figure 3 for an illustration. Numbers in parentheses () refer to item numbers on the illustration.

- Before disassembly, wipe exterior surfaces with a clean cloth to remove dirt, dust and other foreign materials.
- B. When disassembling LED Landing/Taxi/Turn/Engine-Wing Scan/Logo Light (refer to IPL Fig.3 page 906 and 907 for more information) unscrew Bezel (2) from housing (5) using TAE528-100 Spanning Wrench. This will allow the Lens (1), Optics Assembly (3) and Gasket (10) to be removed.
- C. To remove the Power Supply Assy. (6) remove Screws (17) and Cap (7). Then remove Screws (17) to allow the Power Supply Assy. (6), Heat Thermal, PWS (9), Cap (7), and Back Gasket (11) to be removed.
- D. To remove LED Board (4) remove Screws (16) and pull up LED Board (4) and Heat Thermal (8).
- E. To remove Harness (12) from Brass Standoff (20) remove Locknuts (21). Pull Shoulder Washer (18) and flat washer (19) from Standoff (20).
- F. To remove Key (13) remove Screw (15)

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CLEANING

1. Notes.

CAUTION: OBSERVE STANDARD ESD PROCEDURES WHILE HANDLING THE LED LIGHTS IN A DISASSEMBLED STATE.

- A. This section outlines the proper method of cleaning the LED Landing / Taxi / Turn / Scan / Logo Lights.
- A. Refer to the Tables on page 903, 905, and 907 as required for a list of materials needed for assembly (including storage). Equivalent substitutes are acceptable.

2. Cleaning Procedures.

NOTE: Refer to IPL Figure 1, 2, or 3 as required for an illustration. Numbers in parentheses () refer to item numbers on the illustration.

- A. Clean metal parts in Cleaning Compound. Use a cleaning brush to remove ground-in deposits. Use dry cloth to remove all excess Cleaning Compound.
- B. Clean lens (3) with Plastic Cleaner and a clean cloth. Use dry cloth to remove all excess cleaning material.

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TALON AEROSPACE COMPONENT MAINTENANCE MANUAL LANDING / TAXI/TAKE-OFF/TAXI / TURN / SCAN / LOGO LIGHTS

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REPAIR- LED Landing Light TAE0728-1 and Taxi/Take-Off Light TAE0728-1D

1. Notes.

CAUTION: OBSERVE STANDARD ESD PROCEDURES WHILE HANDLING THE LED LANDING AND TAXI/TAKE-OFF LIGHT IN A DISASSEMBLED STATE.

NOTE: Refer to IPL Figure 1 for an illustration. Numbers in parentheses () refer to item numbers on the illustration.

- A. The LED Landing Light TAE0728-1 and Taxi/Take-Off Light TAE0728-1D repair should be limited to the removal and replacement of failed or damaged parts.
- B. Parts can be replaced using the Disassembly instructions provided on page 301.
- C. For overhaul of the LED Landing and Taxi/Take-Off Light contact a TALON AEROSPACE representative at the following address:

TALON AEROSPACE (4FSN4)
Product Support
233 Cahaba Valley Parkway.
Pelham, Alabama 35124
Telephone: (205) 403-6589
FAX: (205) 403-6590
support@talonaerospace.com

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REPAIR- LED Landing / Taxi Light TAE0628-1 and TAE0628-2

1. Notes.

CAUTION: OBSERVE STANDARD ESD PROCEDURES WHILE HANDLING THE LED LANDING / TAXI LIGHT IN A DISASSEMBLED STATE.

NOTE: Refer to IPL Figure 2 for an illustration. Numbers in parentheses () refer to item numbers on the illustration Parts List Reference Chart.

- A. LED Landing / Taxi light repair should be limited to the removal and replacement of failed or damaged parts.
- B. Parts can be replaced using the Disassembly instructions provided on page 302.
- C. For parts required during overhaul of Landing / Taxi Light contact a TALON AEROSPACE representative at the following address:

TALON AEROSPACE (4FSN4)
Product Support
233 Cahaba Valley Parkway.
Pelham, Alabama 35124
Telephone: (205) 403-6589
FAX: (205) 403-6590

support@talonaerospace.com

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REPAIR- LED Landing / Taxi / Turn / Engine-Wing Scan / Logo Light TAE0528-1, TAE0528-1X, TAE0528-1W, TAE0528-2 and TAE0528-2W

1. Notes.

CAUTION: OBSERVE STANDARD ESD PROCEDURES WHILE HANDLING THE LED LANDING, TAXI, TURN, SCAN AND LOGO LIGHT IN A DISASSEMBLED STATE.

NOTE: Refer to IPL Figure 3 for an illustration. Numbers in parentheses () refer to item numbers on the illustration Parts List Reference Chart.

- A. The LED Landing, Taxi, Turn, Engine-Wing Scan, Logo light repair should be limited to the removal and replacement of failed or damaged parts.
- B. Parts can be replaced using the Disassembly instructions provided on page 303.
- C. For parts required during overhaul of Landing, Taxi, Turn, Engine-Wing Scan and Logo Light contact a TALON AEROSPACE representative at the following address:

TALON AEROSPACE (4FSN4)
Product Support
233 Cahaba Valley Parkway.
Pelham, Alabama 35124
Telephone: (205) 403-6589
FAX: (205) 403-6590
support@talonaerospace.com

ASSEMBLY- LED Landing Light TAE0728-1 and Taxi/Take-Off Light TAE0728-1D

1. Notes.

CAUTION: OBSERVE STANDARD ESD PROCEDURES WHILE HANDLING THE LED LANDING AND TAXI/TAKE-OFF LIGHT ASSEMBLY IN A DISASSEMBLED STATE.

- A. This section outlines the proper method for assembly and storing the LED Landing and Taxi/Take-Off Light Assembly.
- B. Examine the parts for damage.
- C. Repair damaged parts before assembly. Refer to REPAIR, section 601, for details.
- D. Refer to the Tables on pages 801 and 802 for a list of materials needed for assembly (including storage). Equivalent substitutes are acceptable.

2. Assembly of Landing and Taxi/Take-Off Light Assembly.

NOTE: Refer to IPL Figure 1 page 902 for an illustration. Numbers in parentheses () refer to item numbers on the illustration. Use Thread locker compound on all screws.

- A. Place Heat Thermal (9) and LED Board (4) into Heat Sink (5). Torque Screws (14) to 2.4 IN.LB.
- B. Install Optics Assembly (3) onto LED Board (4), gently inserting the feet into corresponding holes. Then Place Gasket (11) and Lens (1) onto Heat Sink and screw Bezel (2) onto Heat Sink (5) using TAE728-100 Spanning Wrench to 12 IN.LB. Use anti-seize compound on Heatsink and Bezel threads to avoid binding.
- C. Solder Harness (13) into Power Supply Assembly (6) via solder cups in accordance with IPC J-STD-001. Solder must meet ANSI/J-STD-006.
- D. Place Heat Thermal, PWS (10) into Heat Sink (5), then install Power Supply Assembly (6) into Heat Sink (5), gently sliding it over the pins of LED Board (4) to make a secure connection. Screw Power Supply Assembly (6) into Heat Sink (5) using 3x Screws (15) tightened to 2.4 IN.LB.
- E. Place Back Gasket (12) on Heat Sink (5), then Assemble Harness (13) to Brass Standoffs (18) using Locknut (19) and Nylon Washers (20) and (21) through Cap (7) as shown in Figure 1. Tighten Locknuts to 10 IN.LB. For Landing Light TAE0728-1 there is no polarity and either lead can be mounted to either standoff. For Taxi/Take-Off Light TAE0728-1D polarity is important. Power Supply Assembly (6) is marked with H, L, N. Ensure that each lead is attached to the Brass Standoff (18) which is marked the with same letter on Back Cap (7). The marking on the Back Cap (7) is to be facing out after assembly.
- F. Attach Cap (7) to Heat Sink (5) using Screws (16) torqued to 8 IN. LB. Attach Data Plate (14) with self-adhesive backing and Attach Screws (22) and Washers (23) to Brass Standoffs (18) with 2 IN.LB.
- G. Purge unit and then fill with Nitrogen through Key hole in Heat Sink (5) and attach Key (8) to Heat Sink (5) with Screw (17). Use RTV3145 to seal the unit in case of small leaks around screws.

3. Storage of LED Landing and Taxi/Take-Off Light Assembly.

- A. Before storing, wipe exterior surfaces with a clean cloth to remove dirt, dust and other foreign materials.
- B. Preserve the LED Landing Light Assembly in accordance with Method 44 of MIL-STD-2073.
- C. Place LED Landing and Taxi/Take-Off Light Assembly in a plastic bag with ESD protection.

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TALON AEROSPACE COMPONENT MAINTENANCE MANUAL

LANDING / TAXI/TAKE-OFF/TAXI / TURN / SCAN / LOGO LIGHTS

- D. Place the LED Landing and Taxi/Tak-Off Light Assembly in the unit container with cushion and dunnage sufficiently above, below and around the Landing Light Assembly.
- E. Mark the outside of the unit container in accordance with MIL-STD-129. Include repair dates and location.

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ASSEMBLY- LED Landing / Taxi Light TAE0628-1 and TAE0628-2

1. Notes.

CAUTION: OBSERVE STANDARD ESD PROCEDURES WHILE HANDLING THE LED LANDING / TAXI LIGHT ASSEMBLY IN A DISASSEMBLED STATE.

- A. This section outlines the proper method for assembly and storing the LED Landing / Taxi Light Assembly.
- B. Examine the parts for damage.
- C. Repair damaged parts before assembly. Refer to REPAIR, section 602, for details.
- D. Refer to the Tables on pages 801 and 802 for a list of materials needed for assembly (including storage). Equivalent substitutes are acceptable.

2. Assembly of Landing Light Assembly.

NOTE: Refer to IPL Figure 2 page 904 for an illustration. Numbers in parentheses () refer to item numbers on the illustration. Use Thread locker compound on all screws.

- A. Place Heat Thermal (9) and LED Board (4) into Heat Sink (5). Torque Screws (15) to 2.4 IN.LB.
- B. Install Optics Assembly (3) onto LED Board (4), gently inserting the feet into corresponding holes. Then place Lens (1) and Gasket (10) onto Heat Sink (5) and screw Bezel (2) onto Heat Sink (5) using TAE628-100 Spanning Wrench to 10 IN.LB. Use anti-seize compound on Heatsink and Bezel threads to avoid binding.
- C. Solder Harness (13) into Power Supply Assembly (6) via solder cups in accordance with IPC J-STD-001. Solder must meet ANSI/J-STD-006.
- D. Place Heat Thermal, PWS (12) into Heat Sink (5), then install Power Supply Assembly (6) into Heat Sink (5), gently sliding it over the pins of LED Board (4) to make a secure connection. Screw Power Supply Assembly (6) into Heat Sink (5) using 3x Screws (15) tightened to 2.4 IN.LB.
- E. Place Back Gasket (11) on Heat Sink (5), then assemble Harness (13) to Brass Standoffs (21) using Locknut (18) and Nylon Washers (19) and (20) and Rubber Washers (24) through Cap (7) as shown in Figure 1. Tighten Locknuts to 10 IN.LB.
- F. Attach Cap (7) to Heat Sink (5) using Screws (15) torqued to 8 IN. LB. Attach Data Plate (13) with self-adhesive backing and Attach Screws (22) and Washers (23) to Brass Standoffs (21) with 2 IN.LB.
- H. Purge unit and then fill with Nitrogen through Key hole in Heat Sink (5) and attach Key (8) to Heat Sink (5) with Screw (17). Use RTV3145 to seal the unit in case of small leaks around screws.

3. Storage of LED Landing / Taxi Light Assembly.

- A. Before storing, wipe exterior surfaces with a clean cloth to remove dirt, dust and other foreign materials.
- B. Preserve the LED Landing Light Assembly in accordance with Method 44 of MIL-STD-2073.
- C. Place LED Landing Light Assembly in a plastic bag with ESD protection.
- D. Place the LED Landing Light Assembly in the unit container with cushion and dunnage sufficiently above, below and around the Landing Light Assembly.
- E. Mark the outside of the unit container in accordance with MIL-STD-129. Include repair dates and location.

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ASSEMBLY- LED Landing / Taxi / Turn / Engine-Wing Scan / Logo Light TAE0528-1, TAE0528-1X. TAE0528-1W. TAE0528-2 and TAE0528-2W

1. Notes.

CAUTION: OBSERVE STANDARD ESD PROCEDURES WHILE HANDLING THE LED LANDING / TAXI / TURN / SCAN / LOGO LIGHT ASSEMBLY IN A DISASSEMBLED STATE.

- A. This section outlines the proper method for assembly and storing the LED Landing / Taxi / Turn / Engine-Wing Scan / Logo Light.
- B. Examine the parts for damage.
- C. Repair damaged parts before assembly, Refer to REPAIR, section 603, for details,
- D. Refer to the Tables on pages 801 and 802 for a list of materials needed for assembly (including storage). Equivalent substitutes are acceptable.

2. Assembly of LED Landing / Taxi / Turn / Engine-Wing Scan / Logo Light Assembly.

NOTE: Refer to IPL Figure 3 page 906 for an illustration. Numbers in parentheses () refer to item numbers on the illustration. Use Thread locker compound on all screws.

- A. Place Heat Thermal (8) and LED Board (4) into Heat Sink (5). Torque Screws (16) to 2.4 IN.LB.
- B. Install Optics Assembly (3) onto LED Board (4), gently inserting the feet into corresponding holes. Then Place Gasket (10) and Lens (1) onto Heat Sink (5) and screw Bezel (2) onto Heat Sink (5) using TAE528-100 Spanning Wrench to 10 IN.LB. Use anti-seize compound on Heatsink and Bezel threads to avoid binding.
- C. Solder Harness (12) into Power Supply Assembly (6) via solder cups in accordance with IPC J-STD-001. Solder must meet ANSI/J-STD-006.
- D. Place Heat Thermal, PWS (9) into Heat Sink (5), then install Power Supply Assembly (6) into Heat Sink (5), gently sliding it over the pins of LED Board (4) to make a secure connection. Screw Power Supply Assembly (6) into Heat Sink (5) using 2x Screws (17) tightened to 2.4 IN.LB.
- E. Place Back Gasket (11) on Heat Sink (5), then Assemble Harness (12) to Brass Standoffs (20) using Locknut (21) and Nylon Washers (18) and (19) and Rubber Washers (24) through Cap (7) as shown in Figure 1. Tighten Locknuts (21) to 10 IN.LB.
- F. Attach Cap (7) to Heat Sink (5) using Screws (17) torqued to 8 IN. LB. Attach Data Plate (14) with self-adhesive backing and Attach Screws (22) and Washers (23) to Brass Standoffs (20) with 2 IN.LB.
- I. Purge unit and then fill with Nitrogen through Key hole in Heat Sink (5) and attach Key (13) to Heat Sink (5) with Screw (15). Use RTV3145 to seal the unit in case of small leaks around screws.

3. Storage of LED Landing / Taxi / Turn / Engine-Wing Scan / Logo Light Assembly.

- A. Before storing, wipe exterior surfaces with a clean cloth to remove dirt, dust and other foreign materials.
- B. Preserve the LED Landing / Taxi / Turn / Engine-Wing Scan / Logo Light Assembly in accordance with Method 44 of MIL-STD-2073.

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ASSEMBLY-Landing / Taxi / Turn / Engine-Wing Scan / Logo Light TAE0528-1, TAE0528-1X, TAE0528-1W, TAE0528-2 and TAE0528-2W

- C. Place LED Landing / Taxi / Turn / Engine-Wing Scan / Logo Light Assembly in a plastic bag with ESD protection. Place the LED Landing / Taxi / Turn / Engine-Wing Scan / Logo Light Assembly in the unit container with cushion and dunnage sufficiently above, below and around the Taxi / Turn / Scan / Logo Light Assembly.
- D. Mark the outside of the unit container in accordance with MIL-STD-129. Include repair dates and location.

SPECIAL TOOLS, FIXTURES AND EQUIPMENT

1. Notes.

- A. This section outlines the proper information for special tools, fixtures and equipment used for TESTING, DISASSEMBLY, CLEANING, CHECKS, REPAIR and ASSEMBLY of the LED Light Assembly Components.
- B. Refer to the table below for a list of Special Tools, Fixtures and equipment. Equivalent substitutes may be used.

Table 901. Special Tools, Fixtures and Equipment List

NOMENCLATURE	PART/MODEL NUMBER	USED FOR	FIGURE NUMBER
Variable Regulated Power Supply (0 to 130 VAC, 400 Hz, 3A min)	Model 1500 Ectron Corp. San Diego, CA (24856)	Testing and Fault Isolation	NA
	Or equivalent		
Multimeter (AC and DC Voltage/Ohms/Amps)	Model Fluke 45 Fluke Corporation 6920 Seaway Blvd Everett, WA 98203, USA Or equivalent	Testing and Fault Isolation	NA
Bench Test Unit	Model TAE728TU Talon Aerospace 233 Cahaba Valley Pkwy Pelham, Al 35124	Testing and Fault Isolation	NA
Common Hand tools (screwdrivers, pliers, wrenches, etc.)	Various	Assembly and disassembly	NA
Torque Screw Driver, Inch Lbs. Kit 2-36 Inch Lbs.	Model 304T0036 Sturtevant Richmond 555 Kimberly Drive Carol Stream, IL 60188 Or equivalent	Torque screws where noted during assembly stage	NA
Spanning Wrenches	Model TAE728-100 Model TAE628-100 Model TAE528-100 Talon Aerospace 233 Cahaba Valley Pkwy Pelham, Al 35124	Assembly and disassembly	NA
Hold Down Fixtures	Model TAE728HD Model TAE628HD Model TAE528HD Talon Aerospace 233 Cahaba Valley Pkwy Pelham, Al 35124	Assembly and disassembly	NA

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MATERIAL LIST

Refer to the table below for the authorized List of Materials. Equivalent substitutes may be used.

Table 802. Material List

NOMENCLATURE	SPECIFICATION	MANUFACTURER
Cleaning Cloth	MIL-C-35043	Commercially Available
Plastic Cleaner	MIL-C-18767A	Commercially Available
Cleaning Compound	MIL-DTL-5541	Commercially Available
ESD Bag, Plastic	MIL-PRF-81705-D-Type I (ESD/EMI Opaque Static Shielding)	Commercially Available
Cushion and Dunnage	PPP-C-1752, Unicellular, Polyethylene Foam	Commercially Available
Adhesive Tape	PPP-T-60	Commercially Available
Thread locker	Mil-S-46163A Loctite 222MS	Commercially Available
RTV Coating	MIL-A-46146 Dow Corning RTV-3145	Commercially Available
Anti-Seize Compound	Loctite C5A Copper Anti Seize	Commercially Available

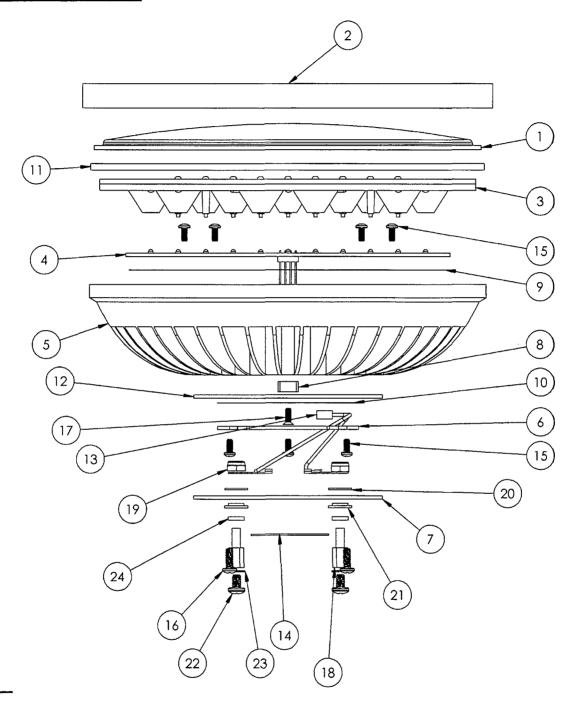


1. Notes.

- A. This section provides an illustrated breakdown of components making up the LED Landing / Taxi / Turn / Engine-Wing Scan / Logo Light Systems. All parts are listed, except for parts which lose their identities by being permanently fastened to other parts or are part of an assembly not subject to disassembly.
- B. Find part numbers by locating the part on the illustration (note the item number). Locate the item number on the Illustrated Parts List. The corresponding part number will be shown on the same line.

2. Reference Chart Column Definitions.

- A. Figure: Refers to the illustration number.
- B. Item: Refers to the item number identifying individual parts on the corresponding Illustrated Parts List Illustration as indicated by the figure number.
- C. Part Number: Defines the manufacturer part number for each illustrated item.
- D. Nomenclature: Identifies parts by name.
- E. Units per Assembly: Defines the quantity of each part listed needed to fully complete assembly of the anticollision light.



LED Landing Light PAR 64 TAE0728-1 and Taxi/Take-Off Liight TAE0728-1D

Figure 1

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ILLUSTRATED PARTS LISTING (SEE FIGURE 1 ON PAGE 902)

Fig Item	PART NUMBER	Nomenclature	Units per Assembly
1- 1	728-001-1	LENS	1
2	728-002-1	BEZEL	1
3	728-003-1	OPTICS ASSEMBLY	1
4	728-004-1	LED BOARD	1
5	728-005-1	HEAT SINK	1
6	728-006-1/728-006-2 *	POWER SUPPLY ASSEMBLY	1
7	728-007-1/728-007-2 **	CAP	1
8	628-008-1	KEY	1
9	728-009-1	HEAT THERMAL	1
10	728-010-1	HEAT THERMAL, PWS	1
11	728-011-1	GASKET	1
12	728-014-1	BACK GASKET	1
13	728-015-1/728-015-2 ***	HARNESS	1
14	DP022/DP034 ****	DATA PLATE	1
15	MS 51957-13	SCREW, PH, #4-40 .250	9
16	MS51957-28	SCREW, PH, #6-32 X .375	4
17	MS24693-C3	SCREW, FH, #4-40 X .313	1
18	8732 ****	#8-32 X.375 BRASS STANDOFF	2 or 3
19	AN 364-832A ****	#8 THIN LOCK NUT	2 or 3
20	5610-503-31 *****	#8 ROUND FLAT WASHER	2 or 3
21	5607-93 *****	#8 SHOULDER WASHER	2 or 3
22	MS51957-41 *****	SCREW, PH, #8-32 X .25	2 or 3
23	MS35333-38 *****	#8 STAR LOCK WASHER	2 or 3

^{*} For TAE0728-1 use Power Supply 728-006-1 and for TAE0728-1D use Power Supply 728-006-2

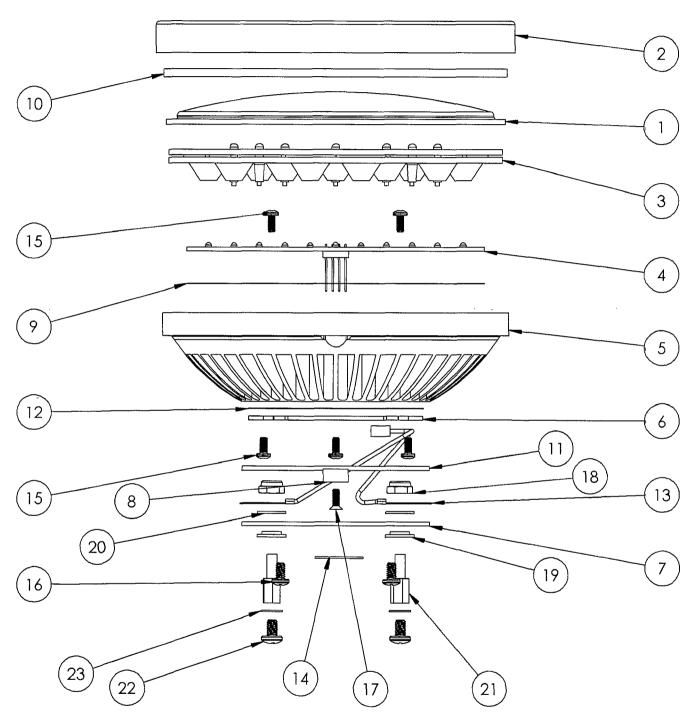
^{**} For TAE0728-1 use Back Cap 728-007-1 and for TAE0728-1D use Back Cap 728-007-2

^{***} Note for TAE0728-1 use Harness 728-015-1 and for TAE0728-1D use Harness 728-015-2

^{****} Note For TAE0728-1 use Data Plate DP022 and for TAE0728-1D use Data Plate DP034

^{*****} For TAE0728-1 use 2 ea of item 18-23, for TAE0728-1D use 3 ea

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LED Landing / Taxi PAR 46 TAE0628-1 and TAE0628-2

Figure 2

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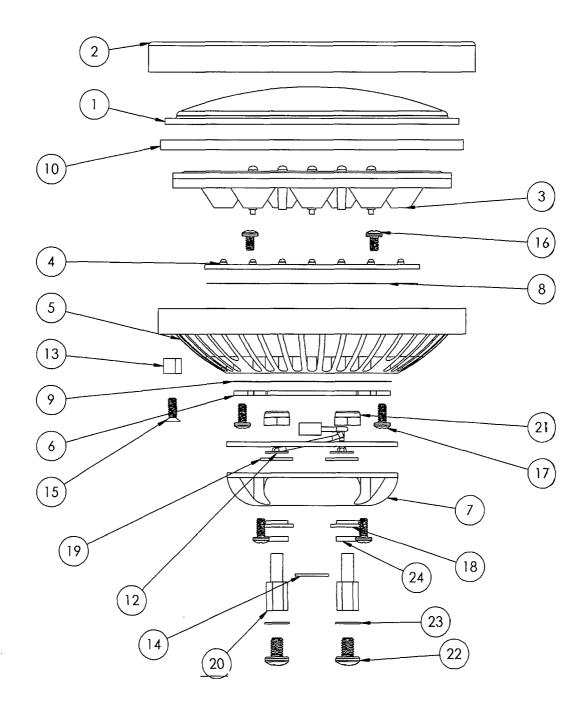
ILLUSTRATED PARTS LISTING (SEE FIGURE 2 ON PAGE 904)

Fig Item	PART NUMBER	Nomenclature	Units per Assembly
2- 1	628-001-1	LENS	1
2	628-002-1	BEZEL	1
3	628-003-1/628-003-2	OPTICAL ASSEMBLY	1
4	628-004-1	LED BOARD	1
5	628-005-1	HEAT SINK	1
6	628-006-1	POWER SUPPLY ASSEMBLY	1
7	628-007-1	BACK CAP	1
8	628-008-1	KEY	1
9	628-009-1	HEAT THERMAL	1
10	628-011-1	GASKET	1
11	628-014-1	BACK GASKET	1
12	728-010-1	HEAT THERMAL, PWS	1
13	728-015-1	HARNESS	1
14	DP020/DP021	DATA PLATE	1
15	MS 51957-13	SCREW, PH, #4-40 .250	7
16	MS51957-26	SCREW, PH, #6-32 X .250	4
17	MS24693-C3	SCREW, FH, #4-40 X .313	1
18	AN 364-832A	#8 THIN LOCK NUT	2
19	5607-93	#8 SHOULDER WASHER	2
20	5610-503-31	#8 ROUND FLAT WASHER	2
21	8732	#8-32 X.375 BRASS STANDOFF	2
22	MS51957-41	SCREW, PH, #8-32 X .25	2
23	MS35333-38	#8 STAR LOCK WASHER	2

^{*} Note: For TAE0628-1 use optics assembly 628-003-1 and for TAE0628-2 use optics assembly 628-003-2

^{**}Note: For TAE0628-1 use Data Plate DP020 and for TAE0628-2 use Data Plate DP021

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LED Taxi / Turn / Engine-Wing Scan / Logo Light PAR 36 TAE0528-1, TAE0528-1W, TAE0528-2 and TAE0528-2W

Figure 3

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Fig Item	PART NUMBER	Nomenclature	Units per Assembly
3- 1	528-001-1	LENS	1
2	528-002-1	BEZEL	1
3	528-003-1/528-003-2 *	OPTICS ASSEMBLY	1
4	528-004-1/528-004-2 528-004-1X **	LED BOARD ASSEMBLY WARM	1
5	528-005-1	HEAT SINK	1
6	528-006-1	POWER SUPPLY ASSEMBLY	1
7	528-007-1	CAP	1
8	528-009-1	HEAT THERMAL	1
9	528-010-1	HEAT THERMAL, PWS	1
10	528-011-1	GASKET	1
11	528-014-1	BACK GASKET	1
12	528-015-1	HARNESS ASSEMBLY	1
13	628-008-1	KEY	1
14	DP016/DP017/DP018/DP019/DP033 ***	DATA PLATE	1
15	MS24693-C3	SCREW, FH, #4-40 X .313	1
16	MS51957-12	SCREW, PH, #4-40 X .188	4
17	MS 51957-13	SCREW, PH, #4-40 .250	6
18	5607-93	#8 SHOULDER WASHER	2
19	5610-503-31	#8 ROUND FLAT WASHER	2
20	8732	#8-32 X.375 BRASS STANDOFF	2
21	AN 364-832A	#8 THIN LOCK NUT	2
22	MS51957-41	SCREW, PH, #8-32 X .25	2
23	MS35333-38	#8 STAR LOCK WASHER	2

^{*} Note: For TAE0528-1 and TAE0528-1W use Optics assembly 528-003-1 and for TAE0528-2 and TAE0528-2W use Optics assembly 528-003-2

33-42-02

^{**} Note: For TAE0528-1 and TAE0528-2 use LED Board assembly 528-004-1 and for TAE0528-1W and TAE0528-2W use LED Board assembly 528-004-2. For TAE0528-1X use LED Board assembly 528-004-1X

^{***}Note: For TAE0528-1 use Data Plate DP016, for TAE0528-1W use Data Plate DP017, for TAE0528-2 use Data Plate DP018 and for TAE0528-2W use Data Plate DP019.

	,	

Page 1 of 5



Electrical Load Statement

For

Talon Aerospace LED Replacement General Purpose Lighting

Project Number ST14465AT-T

Prepared By: Key Marke	Date: 4/14/16
Checked by: Type Hodrson	Date: 4/14/16
Approved by:	

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Revision: 3 Page 2 of 5

RECORD OF REVISIONS

Rev	Date	Pages Affected	Description Of Revision	Approved By
	,			Prepared by: Ken Martin
0	3/18/14	All	Initial Release	Checked by: Tyler Henderson
				Approved by: Jose Perez
				Prepared by: Ken Martin
1	1 8/11/2015 5	5	Added part numbers TAE0428-1, -2, -1W and -2W.	Checked by: Tyler Henderson
				Approved by: Jose Perez
				Prepared by: Ken Martin
2	2 1/19/16 5	5	Removed lamp replaced part numbers with PAR type.	Checked by: Tyler Henderson
			Approved by: Jose Perez	
;				Prepared by: Ken Martin
3	See Front Page	5	1.2 Revised Table.	Checked by: Tyler Henderson
, age		,		Approved by: Jose Perez

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Document No: ELA-728-01

TALON AEROSPACE

Revision: 3 Page 3 of 5

TABLE OF CONTENTS

1.1	Purpose
	Loads

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Document No: ELA-728-01

TALON AEROSPACE Revision: 3 Page 4 of 5

1.1 Purpose

This Electrical Load Statement identifies electrical loads that have been removed from the aircraft as the result of an alteration. Only the rated loads of the individual lamps are recorded in the load charts. The listed loads are based on the value published in the manufacturer's datasheets. The intent of this load statement is to show compliance with CFR part §25.1351(a) (1) (2), Amdt. 25-72

1.2 Loads

Talon Aerospace Part	Equivalent	Power	Lamp Replaced	Power	Difference
Number	Size	Consumption		Consumption	
TAE0728-1	PAR 64	104 Watts	PAR 64	650 Watts	-546
TAE0728-1D High	PAR 64	104 Watts	PAR 64	600 Watts	-496
TAE0728-1D Low	PAR 64	28 Watts	PAR 64	400 Watts	-372
TAE0628-X *	PAR 46	83 Watts	PAR 46	450 Watts	-367
TAE0628-X *	PAR 46	83 Watts	PAR 46	250 Watts	-167
TAE0528-X **	PAR 36	28 Watts	PAR 36	250 Watts	-222
TAE0528-X **	PAR 36	28 Watts	PAR 36	150 Watts	-122
TAE0528-X	PAR 36	28 Watts	PAR 36	100 Watts	-72
TAE0428-X ***	PAR 20	14 Watts	HLX64621	100 Watts	-86
K428-A3-LGL-001	N/A	28 Watts	HLX64621 (2)	200 Watts	-172
K428-A3-LGL-001W	N/A	28 Watts	HLX64621 (2)	200 Watts	-172

^{*=} TAE0628-1, TAE0628-2

1.3 Conclusion

The power added as a result of this alteration is less than the power used by the OEM lamps.

Therefore the power consumption as a result of this modification is negligible to the overall aircraft power system.

^{**=} TAE0528-1, TAE0528-1W, TAE0528-1X, TAE0528-2, TAE0528-2W

^{***=} TAE0428-1, TAE0428-1W, TAE0428-2, TAE0428-2W



Structural Substantiation Report

For

Talon Aerospace

Talon Aerospace
LED Replacement
General Purpose Lighting

Project Number ST14465AT-T

Prepared By: Kee Muste Date: 5/3/16

Checked by: //lliam B. Citing & Date: 5/2/16

Approved by: _____ Date: 5/2/16

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Revision: 2 Page 2 of 7

RECORD OF REVISIONS

Rev	Date	Pages Affected	Description Or Revision	Approved By
0	3/18/14	All:	Initial Release	Prepared by: Tim Stacks Checked by: W.B. Cotney, Jr. Approved by: Ken Martin
1	8/20/15	5	Added K428-B3-LGL-001 & -002. K428-B5-LGL-001, -002, -003, and -004, K428-B7-LGL-001, -002, -003, -004.	Prepared by: Ken Martin
2	See Front Page	5	Added TAE0728-1D, TAE0528-1X, K428-A3-LGL- 001 and -001W.	Prepared by: Ken Martin Checked by: W.B. Cotney Jr. Approved by: Jose Perez

Document No: STR-728-01 Revision: 2 Page 3 of 7

TABLE OF CONTENTS

1	INTRODUCTION
1.1	Purpose
1.2	Modification Description
1.3	Abbreviations
1.4	Units
2	STRUCTURAL SUBSTANTIATION
«	STRUCTURAL SUBSTANTIATION
3	CONCLUSION

TALON AEROSPACE

Document No: STR-728-01

Revision: 2 Page 4 of 7

1 Introduction

1.1 Purpose

The purpose of this report is to provide a structural substantiation for the removal and the replacement of the OEM landing, turn, taxi, scan and logo lights with the Talon Aerospace LED replacement lamps.

1.2 Modification Description

The existing OEM landing, turn, taxi, scan and logo lights are removed and replaced with Talon Aerospace parts. There are no structural changes to the aircraft for the various Talon Aerospace LED replacement lamps.

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TALON AEROSPACE

Revision: 2 Page 5 of 7

The following table provides the part numbers and weights of the removed/replaced parts.

Talon Aerospace Part	Equivalent	Weight (lbs.)	Lamp	Weight (lbs.)	Difference
Number	Size		Replaced		
TAE0728-1	PAR64	2.000	Q4559	1.765	0.235
TAE0728-1D	PAR64	2.000	4557	1.765	0.235
TAE0628-1	PAR46	0.990	4580	0.830	0.160
TAE0628-2	PAR46	0.990	Q4554	0.830	0.160
TAE0528-1	PAR36	0.600	Q4631	0.470	0.130
TAE0528-1X	PAR36	0.600	4596	0.470	0.130
TAE0528-2	PAR36	0.600	4626	0.470	0.130
TAE0528-1W	PAR36	0.600	Q4631	0.470	0.130
TAE0528-2W	PAR36	0.600	4626	0.470	0.130
K428-A3-LGL-001	N/A	1.800	4358532 /	3.000	1.200
			31-2581-X		
K428-A3-LGL-001W	N/A	1.800	4358532 /	3.000	1.200
			31-2581-X		
K428-B3-LGL-001	N/A	1.600	81000-37601	2.200	0.600
K428-B3-LGL-002	N/A	1.600	81000-37601	2.200	0.600
K428-B5-LGL-001	N/A	1.600	4322252	2.000	0.400
			4300181		
			4300181A		
K428-B5-LGL-002	N/A	1.600	4322245	2.000	0.400
	1		4300197		
			4300197A		
K428-B5-LGL-003	N/A	1.600	4322275	2.000	0.400
			4300167		
			4318746		
K428-B5-LGL-004	N/A	1.600	4322268	2.000	0.400
			4300174	ţ	
			4318769		
K428-B7-LGL-001	N/A	1.600	4321554	2.000	0.400
K428-B7-LGL-002	N/A	1.600	4321561	2.000	0.400
K428-B7-LGL-003	N/A	1.600	4321577	2.000	0.400
K428-B7-LGL-004	N/A	1.600	4321584	2.000	0.400

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TALON AEROSPACE

Document No: STR-728-01

Revision: 2 Page 6 of 7

1.3 Abbreviations

Abbreviation	Definition
BA	Balance Arm
FAA	Federal Aviation Administration
FAR	Federal Aviation Regulation
OEM	Original Aircraft Manufacturer
STA	Fuselage Station
STC	Supplemental Type Certificate
WT	Weight

1.4 Units

Unless otherwise specified, all weights are given in pounds (lbs) and all moments are given in inch-pounds (in-lbs).

Document No: STR-728-01 Revision: 2

Page 7 of 7

2 Structural Substantiation

Structural substantiation is based on the comparative weights of the existing lights to the replaced lights.

3 Conclusion

Due to the weight differences of all assemblies and identicality to the original attachment schemes, it is concluded that the assemblies and installations analyzed in this report meet or exceed the original design. No further analysis is necessary.



US Department of Transportation Federal Aviation Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020 11/30/2007

Electronic Tracking Number

For FAA Use Only

INSTRUCTIONS: Print or type all entries. See Title 14 CFR §43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a))

Nationality and Registration Mark N372BG					Serial No. 5164									
1. Aircra	ıft	Make	Make Gulfstream Aerospace						Model S GV-SP (G550)			Series		
2. Owne	Name (As shown on registration certification Contrail Aviation LLC									State PA				
						3.	For	r FAA Use (Only					
	Туре							Unit Identif	ication				-	
Repair	Altera			Unit	N N	1ake	€			Mode	el 			Serial No.
		1	AIRFF	RAME					(As d	escribed in l	Item 1	l above) _	_
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			PROF	ELLER										
			APPL	IANCE	Type Manufacturer									
					6	. Co	onf	ormity Stat	ement					
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City A	ppleton 4914			Sta United S	e <u>WI</u> tates	>	X	Certificated F	Repair Station Maintenance Organization G1FR100Y			G1FR100Y		
hav	e been m	ade in	accord	ance with	ion made to the un the requirements to the best of my k	of P	art	entified in ite 43 of the U.	m 5 abov	e and descr	ibed (attachments hereto e information
Extended per 14 CF App. B					nature/Date of Auth	ht	M	ww Co						14 SEP 2016
								for Return						
Adminis	trator of t	he Fe	deral A		s specified below, dministration and		un	nit identified App		was inspe Reje		n the n	nanner pi	rescribed by the
BY	FAA FIt. S Inspector		ards	Mar	ufacturer		M	aintenance (Organizat	ion		Departi	ment of Tran	by Canadian nsport
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Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished (If more space is required, attach additional sheets. Identify with aircraft nationality a	nd registration mark and date work comple	leted.)						
	N372BG	14 SEP 2016						
Nationality and Registration Mark Date Installed Crane Aerospace & Electronics Tire Sensor Assembly STC ST02408SE, issued 2 SEP 2014, in accordance with Master Document List (MDL) No. CCL83-022-01 REV – as listed on Approved Model List (AML) ST02408SE. Reference STC Letter of Authorization (LOA) dated 14 JUL 2016.								
Updated weight and balance as required. See Weight and Balance re Updated Equipment List as required. See Supplement Equipment List								
END								
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Additional Sheets A	Are Attached							

(3)
US Department
of Transportation
Federal Aviation
Adminintending

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020 2/28/2011	Electronic Tracking Number
F	or FAA Use Only

INSTRUCTIONS: Print or type all entries. See Title 14 CFR §43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a)) Nationality and Registration Mark Serial No. 5164 N372BG 1. Aircraft Make Model Series GV-SP(G550) Gulfstream Aerospace Name (As shown on registration certificate) Address (As shown on registration certificate) Address 5 hog Island Rd Contrail Aviation 2. Owner State Pa Philadelphia City 19153-3809 **United States** Zin Country 3. For FAA Use Only 5. Unit Identification 4. Type Repair Alteration Unit Make Serial No. Model (As described in Item 1 above) AIRFRAME \checkmark POWERPLANT **PROPELLER** Type **APPLIANCE** Manufacturer 6. Conformity Statement B. Kind of Agency A. Agency's Name and Address **Gulfstream Aerospace Services Corporation** U. S. Certificated Mechanic Manufacturer 7 Char Drive Foreign Certificated Mechanic C. Certificate No. Westfield City State MA Certificated Repair Station 01085 USA Zip Country Certificated Maintenance Organization LEGR039G I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge. Signature/Date of Authorized Individual Extended range fuel per 14 CFR Part 43 App. B 10/25/2013 Paul Mumford 7. Approval for Return to Service Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is ✓ Approved Rejected Persons Approved by Canadian FAA Flt. Standards Maintenance Organization Manufacturer Department of Transport Inspector BY Other (Specify) Inspection Authorization FAA Designee Repair Station Signature/Date of Authorized Individual Certificate or Designation No. 10/25/2013 John Cournoyer

FAA Form 337 (10-06)

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Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

	•	N372BG	10/25/2013
		Nationality and Registratio	n Mark Date
133A	Modification- Relocation of Third Fi STC ST01822AT-D dated July 31,	ire Bottle, dated January 17, 2013. A 2011.	pproved by
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United States of America

Bepartment of Transportation -- Federal Abiation Administration

Supplemental Type Certificate

Number ST01822AT-D

This cortificate issued to

Gulfstream Aerospace Corporation 500 Gulfstream Road Savannah, GA 31408

certifies that the change in the type design for the following product with the limitations and conditions therefore as specified hereon weets the aircorthiness requirements of Part 25° of the Federal Aviation Regulations. See Type Cortificate Data Sheet A12EA for complete certification basis.

Original Product-Type Certificate Number:

A12EA

Make:

Gulfstream

*See Page 3

Model

GV & GV-SP

Loseription of Type Design Change. Installation of an APU Fire Bottle in accordance with Gulfstream Aerospace Corporation Index List GC514428001 Rev L, dated July 31, 2011 or later FAA approved revision.

FAA accepted Airplane Flight Manual Supplement, GC51442M001, dated April 8, 1999 or later FAA approved revision is required for fully operational configurations (non-spare). For the model GV, Instructions for Continued Airworthiness, GC51442A001, Revision C, dated September 28, 2010 or later FAA accepted revision is required, and must be made available to the operator at the time of installation. For the model GV-SP, Instructions for Continued Airworthiness, CE52442A000, Revision A, dated October 04, 2010 or later FAA accepted revision is required, and must be made available to the operator at the time of installation.

Similations and Conditions:

- 1. The installer must determine whether this design change is compatible with previously approved modifications.
- 2. If the holder agrees to permit another person to use this certificate to alter a product, the holder must give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked or a termination date is otherwise established by the Administrator of the

Late of application: January 19, 1999

Date reissued.

Date of issuance: April 08, 1999

Friteral Aviation Administration.

Date amended: July 31, 2011



By direction of the Administrator

Milchell Cannon (Signature)

ODA Lead Administrator, ODA-511131-CE

Gulfstream Aerospace Corporation

Savannah, Georgia

(Title)

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·				

United States of America

Bepartment of Transportation - Nederal Abiation Administration

Supplemental Type Certificate

(Continuation Sheet)

Number ST01822AT-D

Date Amended: July 31, 2011

Certification Busis:

The Certification Basis for Gulfstream Model GV is defined as:

- (1) Title 14 CFR Part 25 dated February 1, 1965, including Amendments 25-1 through 25-81 as defined by the TCDS A12EA for all areas not affected by the change.
- (2) Title 14 CFR Part 25 dated February 1, 1965, including Amendments 25-1 through 25-98 for the change and all areas affected by the change. The following lists the Federal Aviation Regulations complied with through Amendment Level 25-98:

25.1, 25.29, 25.301, 25.303, 25.305, 25.307, 25.561, 25.601, 25.603, 25.605, 25.609, 25.613, 25.619, 25.625, 25.901, 25.903, 25.1195, 25.1197, 25.1199, 25.1201, 25.1301, 25.1309, 25.1316, 25.1321, 25.1322, 25.1351, 25.1353, 25.1357, 25.1431, 25.1438, 25.1519, 25.1529, 25.1541, 25.1581, 25.1585.

The Certification Basis for Gulfstream Model GV-SP is defined as:

Title 14 CFR Part 25 dated February 1, 1965, including Amendments 25-1 through 25-98 as defined by the TCDS A12EA.

...END...

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US Department

MAJOR REPAIR AND ALTERATION

Form Approved OMB No. 2120-0020 11/30/2007 Electronic Tracking Number

For FAA Use Only

Of Transp Federal A Administ	Aviation	(Airfram	e, P	owerplan	t, Pr	opeller, or	Ap	pliance)		
thereof) f	for instruction	ns and disp	ositio	ntries. See Ti on of this form (49 U.S.C. §	n. This	report is requ	Part ired	43 Appendix by law (49 U.S	B, and AC 43. S.C. §44701). F	9-1 (or subsequent revision failure to report can result in
		Naționality		Registration				Serial No. 5164		
1. Aircra	itt	Make Gulfstrea	am A	verospace				Model	P (G550)	Series
			_	wn on registra	ation d	ertificate)			shown on regis	stration certificate)
2. Owner	r		СО	NTRAIL AVI	ATIO	N LLC	٠,	Pi		LAND RD PA 19153-3809
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Extended range Signature/Date of Authorized Individual fuel per 14 CFR										
						oval for Retu				
				sons specific Aviation Adm			denti	fied in item 5 Approv		d in the manner prescribed cted
	FAA Flt. Sta Inspector	andards		Manufacture	er	Maintena	nce	Organization		ons Approved by Canadian rtment of Transport
	FAA Design	nee	X	Repair Stati	ion	Inspection	n Au	thorization	Other	· (Specify)
Certifica Designa	te or tion No. G 1	FR100Y		Signature/D		Authorized la	ndivi	dual	-	10/08/2011

Weight and balance limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished (If more space is required, attach additional sheets. Identify with sircraft nationality and registration mark and date work completed).

Installed the 3rd Engine Fire Bottle per ASC 115, (Draft), in accordance with **STC ST01822AT-D**, dated 04/08/99, amended 07/31/2011.

Airplane Flight Manual Supplement No. GC51 442M001, dated 04/08/1999, is required for this approval and is inserted into the appropriate Airplane Flight Manual.

Instructions for Continued Airworthiness: Reference Gulfstream Aerospace Corp Instructions for Continued Airworthiness Document No. CE52 442A000, Rev. A, dated 10/04/2010.

Deactivated the BBML System, bagged and stowed associated winng in accordance with the below listed drawings of which are approved by Dennis N. Halle, AR-E-1107, Electrical Systems, (FAA Project No. TA-03-2011-0008) on FAA Form 8100-9, dated 10/06/2011.

Voice Over IP (BBML) GC514300013, Rev. A Lan Server (BBML) GC214290057, Rev. B Wireless LAN (SBB) GC214280202, Rev. A Fax Interface GC514270277, Rev. A Aux CB Panel (Mod) GC213940342, Rev. A Cockpit Switch Panels GC613700325, Rev. B Satcom High Speed Data (Skylink) GC513030087, Rev. A Cabin LAN CE214S30020' Kev. -DRAWING

Installed a Honeywell/EMS HD-710 in accordance with Gulfstream Aerospace STC ST03737AT-D, issued 10/09/09, with the below listed deviations of which are approved by Dennis N. Halle, AR-E-1107, Electrical Systems, (FAA Project No. TA-03-2011-0008) on FAA Form 8100-9, dated 10/06/2011, and Structures Drawing List CE520000100, Rev. -, dated 10/06/11, of which is approved by William T. Riley, AR-E-1067 Structures, (FAA Project No. TA-03-2011-0008) on FAA Form 8100-9, dated 10/06/2011.

Ground Test Plan EMS And Lynksys Routers And HP Media Drive CE21303G092' K64' -Electrical Load Supplement for Gulfstream G550 S/N 5164 CE214693065, Rev. A Ident Strapping GC213240320, Rev. B Satcom (Mod) GC511030299, Rev. A Wire Routing CE214384003' Kev. -REER CB Panel (Mod) CE213310011, Rev. B LAN Server (BBML) (Mod) CE214290019, Rev. A Aux CB Panel (Mod) GC213940342, Rev. A Satcom High Speed Data (HD-710) CE213030175, Rev. -**SASTEM** <u>DNIMARO</u>

CE21303@002-2104' Kev. -

Instructions for Continued Airworthiness: Reference Gulfstream Aerospace Corp. Instructions for Continued Airworthiness Document No. **GC51 303A011**, Rev. --, dated 09/16/09.

Installed the Broad Band Multi Link (BBML) System. accordance with STC ST02796AT-D, dated 01/21/05, amended 10/26/06.

Ground Test Plan EMS And Lynksys Routers And HP Media Drive

Airplane Flight Manual Supplement No. GC51 303M010, dated 04/03/08, (-511 Configuration) is required for this approval and is inserted into the appropriate Airplane Flight Manual.

Instructions for Continued Airworthiness: Reference Gulfatream Aerospace Corp Instructions for Continued Airworthiness Report No. GC51 303A007, Rev. B, dated 07/27/2010. (-511 Confluration)

Attachment to Interior Completion 337 dated 10/08/2011 GV-SP (G550), S/N: 5164, Registration: N372BG

-SP (G550), S/N: 5164, Registration: N372B	G		<u> </u>	•
Installed a new CNX-200 Unit in accordan which is approved by William T. Riley, AR 8100-9, dated 10/06/2011.				
Modified the aft R/H Lavatory Floor Structs Work was performed in accordance with Sapproved by William T. Riley, AR-E-1067 10/06/2011.	Structural Drawing List CE5	20000100, Rev, date	ed 10/06/11, of wi	hich is
Modified the aft Pressure Bulkhead to allo performed in accordance Structural Drawi William T. Riley, AR-E-1067 Structures, (Fand GAC Drawing CE544980010, Rev, Structures, on FAA Form 8100-9, dated 10	ing List CE520000100, Rev FAA Project No. TA-03-201 dated 09/30/11, of which is	-, dated 10/06/11, of v 1-0008) on FAA Form	which is approved 8100-9, dated 10	d by /06/2011,
Installed 680 Door Assist Handle in accor approved by William T. Riley, AR-E-1067 10/06/2011.				
Incorporated the structural modification poof which is approved by Robert L. Caruthe (FAA Project No. TG-01-2010-0001) on Factorian Modified the R/H Galley Soda Rack Asser Drawing CE523120012, Rev. A1, of which	ers, AR-E-1013 Structures, a AA Form 8100-9, dated 08/3 mbly in accordance with GA n are approved by William T	and Gregg Wizniak, Al 30/10 AC Drawing CE523120	R-E-1009 Powerr 012, Rev, and (olant, GAC
TA-03-2011-0008) on FAA Form 8100-9, o	dated 10/08/2011.			
Aircraft wiring diagrams affected by the ab	oove installations are revise	ed to reflect changes in	system design	,
or their interface to newly installed or mod MIL22759/16 and all new multi-conductor	lified systems as appropriat	te. All new single condu		•
Aircraft Weight & Balance was re-calculat				
				
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US Department
of Transportation
Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller or Appliance)

	Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
F	Foi	FAA Use Only

02/24/11 Darrin Long

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation (49 U.S.C. §46301(a)). Nationality and Registration Mark Serial No. **N372BG** 5164 1. Aircraft Make Model Series **GULFSTREAM AEROSPACE GV-SP (G550)** Name (As shown on registration certificate) Address (As shown on registration certificate) **5 HOG ISLAND RD** 2. Owner **PHILADELPHIA** PA **CONTRAIL AVIATION LLC** City: State: 19153-3809 U.S.A. Country: Zip: 3. For FAA Use Only THE DATA IDENTIFIED HEREIN COMPLIES WITH THE APPLICABLE AIRWORTHINESS REQUIREMENTS AND IS APPROVED FOR THE ABOVE DESCRIBED AIRCRAFT, SUBJECT TO A CONFORMITY-INSPECTION BY A PERSON AUTHORIZED IN FAR 43. SECTION 43.7. DATE: $O_{\lambda} - \lambda$ SIGNATURE OF INSPECTOR: DTW-FSDO PAUL REYNOLDS 4. Type 5. Type Serial No. Make Model Repair Alteration Unit X **AIRFRAME** (As described in Item 1 above) **POWERPLANT PROPELLER** Type **APPLIANCE** Manufacturer 6. Conformity Statement A. Agency's Name and Address B. Kind of Agency U.S. Certificated Mechanic Manufacturer PENTASTAR AVIATION, LLC Foreign Certificated Mechanic C. Certificate No. 7002 HIGHLAND ROAD Certificated Repair Station **WATERFORD, MI 48327-1615** CRS BTVR626C Certificated Maintenance Organization D. I certify that the repair and/or alteration made to the unit(s) identified in Item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and the information furnished herein Is true and correct to the best of my knowledge. Signature / Date of Authorized Individual Extended range fuel per 14 CFR Part 43 App. B. 02/24/11 Darrin Long 7. Approval for Return to Service Pursuant to the authority given persons specified below, the unit identified in Item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is \bowtie Approved Rejected Persons Approved by Canadian Department FAA Fit. Standards Manufacturer Maintenance Organization of Transport Inspector BY Other (Specify) **FAA Designee** X Repair Station Inspection Authorization Certificate or Designation No. Signature / Date of Authorized Individual **CRS BTVR626C**

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

Description of Work Accomplished	
(If more space is required, attach additional sheets. Identify with aircraft nat	ionality and registration mark and date work completed.)
Aircraft Registration Number: N372BG	Date Completed: <u>02/24/2011</u>
The following Alteration was accomplished by Pentastar Aviat # Al10986	ion, LLC and details are on file under Work Order .
ANTI-COLLISION LIGHT ASSEMBLY MOD:	
Removed existing bottom Anti-Collision Beacon Light Assembly P/N 01-0790088-15 in accordance with Whelen Model 900 Installation Guide, Form No. 14002B and Pentastar Dwg. # 3 Anti-Collision Beacon Light Operational Test per Gulfstream Collision	088()-series LED Anti-Collision Light Assembly 368, Rev IR, dated 2/24/11. Performed successful
Changes to aircraft's electrical loading are documented on Pendated 02/24/2011.	tastar Electrical Load Summary Report # 11ELS-001,
The aircraft's Weight and Balance and equipment list have be	en revised to reflect this alteration.
Instructions for Continued Airworthiness pertaining to the on-mentioned Whelen Model 90088()—series installation guide.	
FND	
END	
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Additional Sheets Are Attached

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US Department
of Transportation
Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller or Appliance)

Form Approved OMB No. 2120-0020 11/30/2007	Electronic Tracking Number
For	EAA Hea Only

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation (49 U.S.C. §46301(a)). Nationality and Registration Mark Serial No. N372BG 5164 1. Aircraft Make Model Series **GULFSTREAM AEROSPACE** GV-SP (G550) Address (As shown on registration certificate) Name (As shown on registration certificate) Address: 5 HOG ISLAND RD 2. Owner CONTRAIL AVIATION LLC **PHILADELPHIA** PA City: State 19153-3809 Zip: Country: USA 3. For FAA Use Only 4. Type 5. Type Serial No. Make Model Repair Alteration Unit X **AIRFRAME** (As described in Item 1 above) **POWERPLANT PROPELLER APPLIANCE** Manufacturer 6. Conformity Statement A. Agency's Name and Address B. Kind of Agency U.S. Certificated Mechanic Manufacturer PENTASTAR AVIATION, LLC Foreign Certificated Mechanic C. Certificate No. 7002 HIGHLAND ROAD Certificated Repair Station **WATERFORD, MI 48327-1615** CRS BTVR626C Certificated Maintenance Organization D. I certify that the repair and/or alteration made to the unit(s) identified in Item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and the information furnished herein Is true and correct to the best of my knowledge. Signature / Date of Authorized Individual Extended range fuel per 14 CFR Part 43 App. B. Willis Burnette 02/25/2011 7. Approval for Return to Service Pursuant to the authority given persons specified below, the unit identified in Item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is Approved Rejected Persons Approved by Canadian Department FAA Flt. Standards Manufacturer Maintenance Organization of Transport Inspector BY Other (Specify) **FAA Designee** X Repair Station Inspection Authorization Certificate or Designation No. Signature / Date of Authorized Individual Do mille **CRS BTVR626C** Craig McClure 02/25/2011

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Aircraft Registration Number: N372BG

Date Completed: 02/25/2011

The following Alterations were accomplished by Pentastar Aviation, LLC and details are on file under Work Order # AI-10986.

60 Hz Outlet (MOD)

Installed 1 ea. 60Hz 115 Vac outlet in the Aft LH Baggage compartment in the existing BBML radio rack cabinet in accordance with Pentastar Dwg. # 2471, Rev IR, dated 02/25/2011, which is FAA-DER approved by DERY-410062-CE on FAA Form 8110-3, dated 02/25/2011.

Rechargeable Flashlight (MOD)

Relocated existing flashlight charging base from AFT Airstair storage compartment to behind pilot seat in accordance with Pentastar Dwg. # 25213, Rev IR, dated 02/25/2011, which is FAA-DER approved by DERY-410062-CE on FAA Form 8110-3, dated 02/25/2011.

AirCell Axxess II (MOD)

Installed an Aircell ATG-4000 Broadband Transceiver including Configuration Module (ACM), & interfaced to existing Aircell Axxess II system, existing Aircell Call Alerter (ACA) is replaced with Aircell Call Alerter (ACA-II), in accordance with Pentastar Dwg. # 23346, Rev IR, dated 02/25/2011, which is FAA-DER approved by DERY-410062-CE on FAA Form 8110-3, dated 02/25/2011.

The LRU's are located in the Aft LH Baggage compartment in the existing BBML radio rack cabinet.

Forward Air to Ground (ATG) antenna is installed on the belly (Ref FS 199) in accordance with Strong Aero Engineering Dwg. # N372BG-020211-1, Rev N/C, dated 02/02/11, "Aircell ATG Antenna Installation (FWD)", which is FAA-DER approved by DERT-605818-NM on FAA Form 8110-3, dated 02/20/2011. Reference Strong maintenance requirements for continued airworthiness # N372BG-020211, Rev IR, 02/02/11.

Aft Air to Ground (ATG) antenna is installed on the belly (Ref FS 695) in accordance with Strong Aero Engineering Dwg. # N372BG-020211-2, Rev N/C, dated 02/02/11, "Aircell ATG Antenna Installation (AFT)", which is FAA-DER approved by DERT-605818-NM on FAA Form 8110-3, dated 02/20/11.

Performed successful system ground check of the Aircell Axxess II system in accordance with Aircell Installation Manual # D12004, Rev E, dated March 2010, section 4; and ATG-4000 system in accordance with Aircell Installation Manual # D13485, Rev B, dated November 2009, section 4 with no defects noted.

Reference Pentastar Aviation, LLC Instructions for Continued Airworthiness Number 11CA-001, Revision IR, dated 02/23/2011 for the AirCell Axxess II with Broadband system.

Summary

The above listed maintenance requirements for continued airworthiness have been provided to the owner/operator for incorporation into the operator's maintenance manual and inspection program.

The aircraft's weight and balance report and equipment list have been revised to reflect these alterations.

The effect of these alterations on the aircraft's electrical loading are documented on Pentastar Electrical Load Summary Report # 11ELS-001, dated 02/24/2011.

----- END -----

PA11-007 U.S. DEPARTMENT OF TRANSPORTATION DATE FEDERAL AVIATION ADMINISTRATION 2/25/2011 STATEMENT OF COMPLIANCE WITH THE FEDERAL AVIATION REGULATIONS AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION NAME OF APPLICANT MAKE MODEL NO. TYPE (Airplane, Radio, Helicopter, etc.) Gulfstream G-V-SP (G550) Airplane Pentastar Aviation, LLC LIST OF DATA IDENTIFICATION TITLE Drawings: 23346. Pentastar Aviation, LLC - Wiring Diagram No. 23346, Revision IR, dated 2/25/2011, titled: Aircell Axxess II (MOD). 25213. Wiring Diagram No. 25213, Revision IR, dated 2/25/2011, titled: Rechargeable Flashlight (MOD). 2471. Wiring Diagram No. 2471, Revision IR, dated 2/25/2011, titled: 60 Hz Outlet (MOD). Notes: 1) This approval is for electrical engineering design data only and is not alteration approval. 2) The data listed above demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as "Applicable Requirements" 3) Compliance with additional FAA regulations not listed here may be required. 4) This form does not constitute FAA approval of all the engineering design data necessary for substantiation of compliance to necessary requirements for the entire alteration. ---END----PURPOSE OF DATA In support of major alteration for Gulfstream GV-SP (G550), serial number 5164 only. (N372BG) APPLICABLE REQUIREMENTS (List specific sections) 14 CFR Parts: 25.1301 (a)(1-3); 25.1307 (c); 25.1322 (d); 25.1353 (a); 25.1357 (a)(c); 25.1431 (a)(c)(d). CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under Part 183 of the Federal Aviation Regulations, data listed above and on attached sheets numbered when have been examined in accordance with established procedures and found to comply with applicable requirements of the Federal Aviation Regulations. Recommend approval of these data I (We) Therefore Approve these data SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S) **DESIGNATION NUMBERS(S)** CLASSIFICATION(S) Craig E. McClure DERY-410062-CE Systems & Equipment

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	U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION IANCE WITH THE FEDERAL AVIATION REGULATIONS			DATE 02/20/11			
STATEMENT OF COMPL							
MAKE	MODEL NO.		MPONENT IDENTIFICATI ane, Radio, Helicopter, etc.)		APPLICANT		
GULFSTREAM	GV-SP	AIRPLA			PENTASTAR AVIATION, LLC		
002.011.27.111		LIST OF		TENTANTANTON, LEO			
IDENTIFICATION			TITLE				
DRAWING REFS;							
N372BG-020211-1 REV N/C, 02/02/11	"AIRCELL ATG ANT	ENNA IN	STALLATION (FWD)	1			
N372BG-020211-2 REV N/C, 02/02/11	"AIRCELL ATG ANT	ENNA IN	STALLATION (AFT)"				
REPORT REF:							
N372BG-DT01 REV N/C, 02/02/11	"DAMAGE TOLERA ATG ANTENNA DO			LFSTRE	AM GV-SP FWD AIRCELL		
	Notes: This approval is for e	engineerii	ng design data only. It	indicates	s the data listed below		
	demonstrates compliance only with the regulations specified by paragraph and subparagraph listed below as 'APPLICABLE REQUIREMENTS'. This form does not constitute FAA approval of all the engineering data necessary for substantiation of compliance to necessary requirements for the entire alteration/repair. Structural aspects are approved, electrical aspects are not included						
·	The Aircell ATG (FWD) Antenna installation requires a Threshold Inspection and Repeat Inspections. These inspection limitations shall be provided by way of the Airplane Maintenance Manual Supplement for Continued Airworthiness, Report Ref; N372BG-ICAW-020211, Rev IR, 02/02/11						
PURPOSE OF DATA							
IN SUPPORT OF MAJOR A FOR ONE AIRCRAFT ONL	Y; S/N <u>5164,</u> T/N <u>N37</u>		MPLIANCE WITH THE	FOLLO	WING REGULATIONS		
APPLICABLE REQUIREMENTS (List specific sections) 14 CFR Part 25.301(a)(b), Amdt 25-23; .303, Amdt 25-23; .305(a), Amdt 25-86; .307(a), Amdt 25-72; .365(a)(b)(d), Amdt 25-87; .561(a)(b)(3)(c), Amdt 25-91; .571(a)(b), Amdt 25-96; .601, Amdt 25-0; .603(a)(b)(c), Amdt 25-46; .605(a), Amdt 25-46; .607(a), Amdt 25-23; .609(a), Amdt 25-0; .613(a)(b)(c), Amdt 25-112; .625(a)(b), Amdt 25-72				.601, Amdt 25-0;			
CERTIFICATION - Under author Part 183 of the Federal Aviation F established procedures and found	Regulations, data listed abov	e and on at	tached sheets numbered <u>N</u>	V/A have b			
	end approval of these data these data						
SIGNATURE(S) OF DESIGNATED EN	GINEERING REPRESENTATIV	/E(S)	DESIGNATION NUMBER(S)	CLAS	SSIFICATION(S)		
TREVOR J. STRONG	1 Strong	<u></u>	DERT-605818-NM	STF	RUCTURES		
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ELECTRICAL LOAD SUMMARY REPORT

Pentastar Aviation, LLC. 7002 Highland Road Waterford MI. 48327 Report No. 11ELS-001 Reg. No. N372BG Model No. GAC GV-SP Serial No. 5164 Work Order No. AI- 10986

PURPOSE

This Electrical Load Statement identifies electrical loads that have been removed from or added to the aircraft as a result of an alteration. The listed loads are calculated based on the maximum value published in the manufacturer's technical manuals. Only busses listed have been modified by Pentastar Aviation, LLC. (PA) For additional information on electrical loading, refer to the Aircraft's Electrical Load Analysis.

REMOVED LOADS

NO	Item Removed	CB TITLE	BUS TITLE	LOAD	CB Size
1	Bottom Beacon (Normal Operation)	BOT AC LT GND OPER	28VDC GND SVC	3 amp	5 amp
2	Bottom Beacon (Ground Operation)	BOT AC LT	28VDC R EES	3 amp	5 amp

ADDED LOADS

NO	Item Installed	CB TITLE	BUS TITLE	LOAD	CB Size
1	AIRCELL ATG-4000	ATG	28VDC MAIN	5.3 amp	10 amp
2	AIRCELL ACM	ACM	28VDC MAIN	1.0 amp	3 amp
3	Bottom Beacon (Normal Operation)	BOT AC LT GND OPER	28VDC GND SVC	3 amp	5 amp
4	Bottom Beacon (Ground Operation)	BOT AC LT	28VDC R EES	3 amp	5 amp

REVISED LOADS

BUSS	28VDC MAIN (SW)	28VDC GND SVC	28VDC R EES
CHANGE	+6.3 amp	0.0 amp	0.0 amp

INSTRUCTIONS

This summary of changes to the electrical loading is provided to the owner/operator of above identified Aircraft. It is the responsibility of such owner/operator to file a copy of this summary with the aircraft's records along with the existing load analysis documents.

PREPARED BY:

CHECKED BY:

APPROVED BY:

Dave Pemberton

02/24/11

Date

Kurt Diffenbaugh

02/24/11

Date Crai

02/24/11 Date

Original to Design Records

Copy to Work Order

Copy Attach to Customer 337

Copy Attach to FAA 337

Copy Attach to 337 File

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Form Approved OMB No. 2120-0020 11/30/2007 Electronic Tracking Number

US D	epartment		MA	JO	R REPAIR	AN	D A	ALTERATION			For F	AA Use Only		
Of Tra	ansportatio	on (Airframe	, P	owerplant	t, Pr	ope	eller, or Appliar	nce)					
	ral Aviation							,	. 1					
INIST	Inistration	IS: Dri	nt or type s	م اا	tries See Ti	tle 1	4 CF	R & 43.9 Part 43 A	Appendi	x B. and	AC 43.9-	1 (or subsequent revision		
there	of) for ins	truction	ns and dispo	sitio	n of this form 19 U.S.C. §46	. This	s rep	ort is required by law	v (49 U.	S.C. §447	701). Failu	re to report can result in a		
OIVII P	somany to		Nationality	and	Registration			Seri	al No.					
1. Aiı	rcraft		N37	2B(<u> </u>				5164			Corios		
1. Aircraft			Make Gulfstrea					Model GV-SP (G550)			Series			
			Name (As	sho	wn on registra	ation	certii	ficate) Add	ress (As			tion certificate)		
2. Owner				NTRAIL AVI	TRAIL AVIATION LLC			5 HOG ISLAND RD PHILADELPHIA, PA 19153-3809						
	-						3. F	or FAA Use Only						
	4. Type					5	. Un	it Identification						
Repa	air Alte	ration	Unit		Make				Model			Serial No.		
] [X	AIRFRAME					(As desc	(As described in Item 1 above)					
] [POWERPL	ANT										
] [PROPELLE	Ŕ					<u></u>					
					Туре									
] [APPLIANCE	E	Manufacturer					;				
								nformity Statement		-				
A. A	gency's r	Name :	and Addres	S		B. I		of Agency S. Certificated Mechan	nic	· 1	Manufa	cturer		
Gulf	fstream A	Aerosi	pace Corpo	orati				reign Certificated Mec				C. Certificate No.		
	365 Disco							rtificated Repair Statio						
	leton, W					Certificated Mainter			enance			G1FR100Y		
						<u> </u>	<u> </u>	ganization						
ŀ	hereto hav	ve beer	n made in ac	cord	lance with the	e requ	uiren	(s) identified in item the nents of Part 43 of the est of my knowledge.	e U.S. F	and desc ederal A	cribed on t viation Re	he reverse or attachments gulations and that the		
ļ	ended ran							ized Individual						
	per 14 C			•					1/11	118-2-11	,	00/00/0040		
	43 App.		-		Robert J. I	Hellm	ann	Sobut 19	TUK M			03/26/2010		
								al for Return to Ser						
Purs by ti	suant to t	the aut	hority giver or of the fed	n pe eral	rsons specifi Aviation Adn	ed b	elow ratio	n, the unit identified on and is ⊠	in item Appro	5 was in ved l	spected i □ Rejecte	in the manner prescribed		
BY	FAA		andards					Maintenance Orga				s Approved by Canadian nent of Transport		
-	FAA	Desig	nee	x	Repair Stat	ion		Inspection Authoriz	zation		Other (S	Specify)		
Cert	Certificate or Signature/Date of Authorized Individual													
l			1FR100Y		Robert I H	ellma	nn (Polyst a	Allh	USMI	1	03/26/2010		

Weight and balance limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed).

Reworked Credenza Cabinet Installing eight (8) port Ethernet Switch in the Credenza Cabinet in accordance with the following GAC Engineering Electrical Drawings of which are DER approved by by Peter Jobe DERY-830436-CE on FAA 8110-3 form dated 03/25/2010:

DRAWING

SYSTEM

CE51 4290019, Rev.--CE51 4940121, Rev.-- Lan Server (BBML)(Mod)

60 Hz Outlets (Mod)

Reworked Credenza Cabinet to accommodate below listed drawing in accordance with the following electrical GAC Engineering Structural Drawing of which are DER approved by William T Riley DERY-831142-CE on FAA 8110-3 form dated 03/26/2010:

DRAWING

SYSTEM

CE522360061, Rev.--

Lan Server (BBML)(Mod)

Electronic Cable and additional wire used complies with the burn testing requirements of FAR 25.869 (a)(4) Amendment 25-113 Appendix F Part I (a)(3) which is approved by Gary K. Palmer, DERY-605430-CE on FAA 8110-3 form dated 03/26/2010.

Installed Software for Certification Foxtrot (F) in accordance with Gulfstream Engineering Top Drawing No. 1159ASC57908 - Top Drawing ASC 908 Planeview Master Operating System Software Update (Cert Foxtrot) of which is approved under ODA Project No. ODA-T-2009-GEN-SA-001 on FAA 8100-9 form dated 10/29/2009.

is approved under ODA	Project No. ODA-T-2009-GEN-SA-001 on FAA 8100-9 form dated 10/29/2009.									
The Aircraft Weight and	Balance is negligible.	=								
		END				****				
			•							
•	•									

Additional Sheets are attached.

2
US Department
of Transportation Federal Aviation
Administration

Form Approved OMB No. 2120-0020

Electronic Tracking Number

US Department of Transportation Federal Aviation			. (A			AND ALTERATION , Propeller, or Applia	11/30/2007						
		viation	(A)	ir ir aine, i ower p	iant	, rropener, or Appua	ince,	For FAA Use Only					
Administration						·							
INSTRUCTIONS: Print or type all entries. See Title 14 CFR §43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a))													
1	\ircra	.6.	Nationality	and Registration Ma N3	rk 72BC	;	S	Serial No. 5164					
1. 7	XII CI A	iit	Make	Gul	stream			Model GV-SP		Series G550			
			Name (As s	hown on registration	cert	ificate)	A	ddress (As	show	n on registration o	certificate)		
2. Owner				Contrail A	ati	on II C		Address 5 Hog Island Road City Philadelphia State PA					
				Contrait A	viaii	on EEC		ip <u>19153-</u>	intry <u>USA</u>				
	_					3. For FAA Use On	ılv			_	<u> </u>		
							<u> </u>						
		4. Type				5.	Unit I	lentificati	on				
	Re	pair	Alteration	Unit		Make			Mode	ıl	Serial No.		
	[]		AIRFRAME	-	(As describ		Is describe	ed in I	tem I above)			
	D POWERPLANT												
☐ ☐ PROPELLER													
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				<u> </u>	IVI								
Δ	Agenc	v's Name	and Address		В	6. Conformity Statement B. Kind of Agency							
A. Agency's Name and Address Name Gulfstream Aerospace Corporation					U.S. Certified Mechanic				Manufacturer				
Address 500 Gulfstream Road					Foreign Certified Mechanic				Certificate No.				
	ty <u>Sav</u> 314	vannah 08		State <u>GA</u> Country <u>USA</u>	X Certificated Repair Station Certificated Maintenance Organiz			nization	GR4R2167 Specialized SP Series, A	f: Accessory Class: [,]] & [] [/ Radio Services, Airframe: All G-159, G-215 Istra, G100, Galaxy & G200 Series Ai	Class: I, II, III / Instrument Class: I, II & III, NDT 19.G-1159A, O-IV, GIV-SP, GIV-X Series, GV, GV- icrash, Powerplant: RR MK-529, MK511-8, MK611- E731-3A-200G, TFE731-3C-200G a PW306A		
					unit(s) identified in item 5 above							
	accor best o	dance with of my knowl	the requireme ledge.		J.S. F	ederal Aviation Regulations	and tha	at the infor					
	Extended range fuel Signature/Date of Authorized Individual per 14 CFR Part 43												
App. B Charles A. Ries 397 10/08/2009							10/08/2009						
_						. Approval for Return to							
	Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED												
FAA Flt Standards			Manufacturer		Maintenance Organizati	on	-		Persons Approved Department of Tr				
וטו		FAA Des	ignee	X Repair Station		Inspection Authorizations			Other (Specify)				
Cer	tificat	e or		Signature/Date of	Autho	prized Individual							
		on No.				γ_{α}							
GR4R216M Charles A. Ries					lele.		397			10/08/2009			

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished (If more space is required, attach additional sheets, Identify with aircraft nationality and registration mark and date work accomplished.)					
Reference: Gulfstream Aerospace Corporation Work Order No. SC180396					
N372BG 10/08/2009 Nationality and Registration Mark Date					
 Complied with mid cabin credenza modification in accordance with Gulfstream Custom Engineering Drawing List CE52 005164 Rev. NC. Reference FAA Form 8100-9 dated 10/07/2009 to substantiate Drawing CE52 2360058 Rev. NC. 					
(Modification – Fax Drawer, Mid Cabin Credenza).					
Additional Sheets Are Attached					

US Department

Form Approved	Electronic
OMB No. 2120-0020	
11/30/2007	1

Tracking Number

MAJOR REPAIR AND ALTERATION For FAA Use Only (Airframe, Powerplant, Propeller, or Appliance) Of Transportation Federal Aviation Administration INSTRUCTIONS: Print or type all entries. See Title 14 CFR § 43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a)) Serial No. Nationality and Registration Mark N372BG 5164 1. Aircraft Model Series Make **Gulfstream Aerospace GV-SP (G550)** Address (As shown on registration certificate) Name (As shown on registration certificate) 2. Owner **5 GOG ISLAND RD CONTRAIL AVIATION LLC** PHILADELPHIA PA 19153-3809 3. For FAA Use Only 4. Type 5. Unit Identification Alteration Model Serial No. Repair Unit Make X **AIRFRAME** (As described in Item 1 above) **POWERPLANT** \Box **PROPELLER** Type \Box **APPLIANCE** Manufacturer 6. Conformity Statement A. Agency's Name and Address B. Kind of Agency U.S. Certificated Mechanic Manufacturer **Gulfstream Aerospace Corporation** C. Certificate No. Foreign Certificated Mechanic W6365 Discovery Drive Certificated Repair Station **G1FR100Y** Appleton, WI 54914 Certificated Maintenance Organization I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge. Signature/Date of Authorized Individual Extended range fuel per 14 CFR David L. Wellnitz 08/12/08 Part 43 App. B 7. Approval for Return to Service Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the federal Aviation Administration and is ☐ Rejected Persons Approved by Canadian FAA Flt. Standards Manufacturer Maintenance Organization Department of Transport Inspector BY

Inspection Authorization

David S.Wellni

Other (Specify)

08/12/08

Certificate or

FAA Designee

Designation No. G1FR100Y

Repair Station

David L. Wellnitz

Signature/Date of Authorized Individual

Weight and balance limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

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on FAA Form 8110-3 dated 08/12/08.	
w Honeywell WINN Graphic Weather System in accordance with the below listed Drawings of are DER John X. Schober Jr., DERY-410129-CE, on FAA Form 8110-3 dated 08/11/08, and Peter Jobe, DERY-	pproved by
Work Accomplished is additional sheets. Identify with aircraft nationality and registration mark and date work completed).	

UNITED STATES OF AMERICA DEPARTMENT OF TRANSPORTATION—FEDERAL AVIATION ADMINISTRATION SPECIAL AIRWORTHINESS CERTIFICATE PURPOSE TO Show Complaince with the Begulation to Bond MANU- NAME N/A MANU-FACTURER ADDRESS FROM FLIGHT TO N/A N- 372BG MODEL (G550)BUILDER Gulfstream DATE OF ISSUANCE\\\ 1-4-08 **EXPIRY** OPERATING LIMITATIONS DATED ARE A PART OF THIS CERTIFICATE SIGNATURE OF FAA REPRESENTATIVEL RESIGNATION OR OFFICE NO. \D&R#-609001-CE

Any alteration, reproduction or misuse of this certificate may be punishable by a fine not exceeding \$1,000 or imprisonment not exceeding 3 years, or both. THIS GERTIFICATE MUST BE DISPLAYED IN THE AIRCRAFT IN ACCORDANCE WITH APPLICABLE FEDERAL AVIATION REGULATIONS.

Ε

Α	This airworthiness certificate is issued under the authority of the Federal Aviation Act of 1958 and the Federal Aviation Regulations (FAR).
	This airworthiness certificate authorizes the manufacturer named on the reverse side to conduct production flight tests, and only production flight tests, of aircraft registered in his name. No person may conduct production flight tests under this certificate: (1) Carrying persons or property for compensation or hire; and/or (2) Carrying persons not essential to the purpose of the flight.
С	This airworthiness certificate authorizes the flight specified on the reverse side for the purpose shown in Block A:
D	This airworthiness certificate certifies that, as of the date of issuance, the aircraft to which issued has been inspected and found to meet the requirements of the applicable FAR. The aircraft does not meet the requirements of the applicable comprehensive and detailed airworthiness code as provided by Annex 8 to the Convention On International Civil Aviation. No person may operate the aircraft described on the reverse side; (1) except in accordance with the applicable FAR and in accordance with conditions and limitations which may be prescribed by the Administrator as part of this certificate; (2) over any foreign country without the special permission of that country.
Ε	Unless sooner surrendered, suspended, or revoked, this airworthiness certificate is effective for the duration and under the conditions prescribed in FAR Part 21, Section 21.181 or 21.217.



Small Airplane Directorate
Manufacturing Inspection District Office
6020 28th Avenue South, Room 103
Minneapolis, MN 55450-2700

EXPERIMENTAL OPERATING LIMITATIONS RESEARCH AND DEVELOPMENT/SHOWING COMPLIANCE WITH THE REGULATIONS

MAKE: Gulfstream

MODEL: GV-SP (G550) SERIAL NUMBER: 5164

REGISTRATION NUMBER: N372BG

SUPERSEDED DARF-609001-CE

Retain these limitations in the aircraft during each flight.

- 1. No person may operate this aircraft unless FAA Form 8130-7 is displayed at the cabin or cockpit entrances so that it is visible to passengers or flightcrew members.
- 2. No person may operate this aircraft for other than the purpose of **RESEARCH AND DEVELOPMENT** / **SHOWING COMPLIANCE WITH THE REGULATIONS** to accomplish the flight operation outlined in the program letter dated December 18, 2007, which describes compliance with §21.193(d), and has been made available to the pilot in command of the aircraft. In addition, this aircraft must be operated in accordance with applicable air traffic and general operating rules of part 91 and all additional limitations herein prescribed under the provisions of §91.319(i).
- 3. All flights shall be conducted within the geographical area described as follows:
 - 1000 NM radius around the Outagamie County Regional Airport, Appleton, Wisconsin (ATW), excluding Canadian airspace. Except for takeoff and landing operations this aircraft must be operated over open water or in sparsely populated areas having light air traffic.
- 4. When changing between operating purposes of a multiple-purpose certificate (Research and Development and Show Compliance), the operator must determine that the aircraft is in a condition for safe operation and appropriate for the purpose intended. A record entry will be made by an appropriately rated person to document that finding in the aircraft logbook.
- 5. This aircraft must not be operated unless it is inspected and maintained in accordance with appropriate military technical publications and/or manufacturer's recommendations. The owner/operator must select, establish, identify, and use an inspection program as set forth in § 91.409(e), (f), (g), and (h). This inspection program must be recorded in the aircraft maintenance records.
- 6. The pilot in command of this aircraft must hold an appropriate category/class rating. If required for the type of aircraft to be flown, the pilot-in-command also must hold either an appropriate type rating or a letter of authorization issued by a FAA Flight Standards Operations Inspector.
- 7. This aircraft may be operated under VFR, day and/or night.
- 8. This aircraft may be operated under IFR, and must be properly equipped for instrument flight in accordance with § 91.205.
- 9. No person may operate this aircraft for carrying persons or property for compensation or hire.
- 10. No person may be carried in this aircraft during flight unless that person is essential to the purpose of the flight.

 The pilot shall record the names of the persons carried on each flight.
- The pilot in command of this aircraft shall advise each person of the experimental nature of this aircraft, and explain that it does not meet the certification requirements of a standard certificated aircraft.

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- 12. This aircraft must contain the placards, markings, etc., (or other operating instructions developed for an STC modification) required by §91.9.
- 13. This aircraft is prohibited from aerobatic flight; that is, an intentional maneuver involving an abrupt change in the aircraft's attitude, an abnormal attitude, or abnormal acceleration not necessary for normal flight.
- 14. The aircraft must not be used for glider towing, banner towing, or intentional parachute jumping.
- 15. No person may operate this aircraft unless within the preceding 12 calendar months it has had a condition inspection performed in accordance with Appendix D to Part 43, or other FAA-approved programs, and found to be in a condition for safe operation. This inspection will be recorded in the aircraft maintenance records.
- 16. Only FAA-certified mechanics with appropriate rating as authorized by § 43.3 may perform inspections required by these operation limitations.
- 17. Inspections must be recorded in the aircraft maintenance records showing the following or a similarly worded statement: "I certify that this aircraft has been inspected on (insert date) in accordance with the scope and detail of appendix D to part 43, or other FAA-approved programs, and was found to be in a condition for safe operation." The entry will include the aircraft's total time-in-service, and the name, signature, certificate number, and type of certificate held by the person performing the inspection.
- 18. If aircraft or engine operating limits are exceeded, an appropriate entry will be made in the aircraft records.
- 19. This aircraft must not be operated unless it is maintained and inspected in accordance with the requirements of Part 43.
- 20. The aircraft must display the word **EXPERIMENTAL** in accordance with §45.23(b).
- 21. The pilot in command of this aircraft shall notify the air traffic control tower of the experimental nature of this aircraft when operating into or out of airports with operating control towers. The pilot in command shall plan routing that will avoid densely populated areas and congested airways when operating VFR.
- 22. This aircraft does not meet the requirements of the applicable, comprehensive, and detailed airworthiness code as provided by Annex 8 to the Convention on International Civil Aviation. The owner/operator of this aircraft must obtain written permission from another country's CAA prior to operating this aircraft in or over that country. That written permission must be carried aboard the aircraft together with the U.S. airworthiness certificate and, upon request, be made available to an FAA inspector or the CAA in the country of operation.
- 23. Aircraft instruments and equipment installed and used under §91.205 must be inspected and maintained in accordance with the requirements of parts 43 and 91. Any maintenance or inspection of this equipment must be recorded in the aircraft maintenance records.
- 24. Application must be made to the Minneapolis MIDO for any revision to these operating limitations.
- 25. In accordance with §47.45, the FAA Aircraft Registry must be notified within 30 days of any change to the aircraft registrant's address. Such notification is made by submitting a FAA Form 8050-1, Aircraft Registration Application, to AFS-750 in Oklahoma City, Oklahoma.
- 26. The pilot in command shall ensure all passengers carried onboard the aircraft are necessary for the purpose of the intended flight, and will list each passenger by name on an aircraft manifest.

These operating limitations are a part of a special airworthiness certificate issued: January 4, 2008

Date of Limitations Issuance: January 4, 2008

Date of Expiration: February 3, 2008

SUPERISEDED DARF-609001-CE

Emery P. Wiltse

DAR-609001-CE

30-100POL-19-1-109-01-0E

FAA FORM 8130-6, APPLICATION FOR U.S. AIRWORTHINESS CERTIFICATE

Form Approved

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	DEPARTMENT OF TRANSPORTATION—FEDERAL AVIATION ADMINISTRATION SPECIAL AIRWORTHINESS CERTIFICATE	
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E	SIGNATURE OF FAA REPRESENTATIVE () DESIGNATION OR OFFICE NO.	رد. در د
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Any	alteration, reproduction or misuse of this certificate may be punishable by a fine not exceeding \$1.00 risonment not exceeding 3 years or both THIS CERTIFICATE MUST BE DISPLAYED IN THE AIRCRAFT CORDANCE WITH APPLICABLE FEDERAL AVIATION REGULATIONS.)O T

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Small Airplane Directorate Manufacturing Inspection District Office 6020 28th Avenue South, Room 103 Minneapolis, MN 55450-2700

EXPERIMENTAL OPERATING LIMITATIONS RESEARCH AND DEVELOPMENT/SHOWING COMPLIANCE WITH THE REGULATIONS

(

MAKE: Gulfstream

MODEL: GV-SP (G550) SERIAL NUMBER: 5164

REGISTRATION NUMBER: N372BG

Retain these limitations in the aircraft during each flight.

- 1. No person may operate this aircraft unless FAA Form 8130-7 is displayed at the cabin or cockpit entrances so that it is visible to passengers or flightcrew members.
- 2. No person may operate this aircraft for other than the purpose of **RESEARCH AND DEVELOPMENT** / **SHOWING COMPLIANCE WITH THE REGULATIONS** to accomplish the flight operation outlined in the program letter dated December 18, 2007, which describes compliance with §21.193(d), and has been made available to the pilot in command of the aircraft. In addition, this aircraft must be operated in accordance with applicable air traffic and general operating rules of part 91 and all additional limitations herein prescribed under the provisions of §91.319(i).
- 3. All flights shall be conducted within the geographical area described as follows:
 - 1000 NM radius around the Outagamie County Regional Airport, Appleton, Wisconsin (ATW), excluding Canadian airspace. Except for takeoff and landing operations this aircraft must be operated over open water or in sparsely populated areas having light air traffic.
- 4. When changing between operating purposes of a multiple-purpose certificate (**Research and Development and Show Compliance**), the operator must determine that the aircraft is in a condition for safe operation and appropriate for the purpose intended. A record entry will be made by an appropriately rated person to document that finding in the aircraft logbook.
- 5. This aircraft must not be operated unless it is inspected and maintained in accordance with appropriate military technical publications and/or manufacturer's recommendations. The owner/operator must select, establish, identify, and use an inspection program as set forth in § 91.409(e), (f), (g), and (h). This inspection program must be recorded in the aircraft maintenance records.
- 6. The pilot in command of this aircraft must hold an appropriate category/class rating. If required for the type of aircraft to be flown, the pilot-in-command also must hold either an appropriate type rating or a letter of authorization issued by a FAA Flight Standards Operations Inspector.
- 7. This aircraft may be operated under VFR, day and/or night.
- 8. This aircraft may be operated under IFR, and must be properly equipped for instrument flight in accordance with § 91.205.
- 9. No person may operate this aircraft for carrying persons or property for compensation or hire.
- 10. No person may be carried in this aircraft during flight unless that person is essential to the purpose of the flight.

 The pilot shall record the names of the persons carried on each flight.
- The pilot in command of this aircraft shall advise each person of the experimental nature of this aircraft, and explain that it does not meet the certification requirements of a standard certificated aircraft.

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- 12. This aircraft must contain the placards, markings, etc., (or other operating instructions developed for an STC modification) required by §91.9.
- 13. This aircraft is prohibited from aerobatic flight; that is, an intentional maneuver involving an abrupt change in the aircraft's attitude, an abnormal attitude, or abnormal acceleration not necessary for normal flight.
- 14. The aircraft must not be used for glider towing, banner towing, or intentional parachute jumping.
- 15. No person may operate this aircraft unless within the preceding 12 calendar months it has had a condition inspection performed in accordance with Appendix D to Part 43, or other FAA-approved programs, and found to be in a condition for safe operation. This inspection will be recorded in the aircraft maintenance records.
- 16. Only FAA-certified mechanics with appropriate rating as authorized by § 43.3 may perform inspections required by these operation limitations.
- 17. Inspections must be recorded in the aircraft maintenance records showing the following or a similarly worded statement: "I certify that this aircraft has been inspected on (insert date) in accordance with the scope and detail of appendix D to part 43, or other FAA-approved programs, and was found to be in a condition for safe operation." The entry will include the aircraft's total time-in-service, and the name, signature, certificate number, and type of certificate held by the person performing the inspection.
- 18. If aircraft or engine operating limits are exceeded, an appropriate entry will be made in the aircraft records.
- 19. This aircraft must not be operated unless it is maintained and inspected in accordance with the requirements of Part 43.
- 20. The aircraft must display the word **EXPERIMENTAL** in accordance with §45.23(b).
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- 22. This aircraft does not meet the requirements of the applicable, comprehensive, and detailed airworthiness code as provided by Annex 8 to the Convention on International Civil Aviation. The owner/operator of this aircraft must obtain written permission from another country's CAA prior to operating this aircraft in or over that country. That written permission must be carried aboard the aircraft together with the U.S. airworthiness certificate and, upon request, be made available to an FAA inspector or the CAA in the country of operation.
- 23. Aircraft instruments and equipment installed and used under §91.205 must be inspected and maintained in accordance with the requirements of parts 43 and 91. Any maintenance or inspection of this equipment must be recorded in the aircraft maintenance records.
- 24. Application must be made to the Minneapolis MIDO for any revision to these operating limitations.
- 25. In accordance with §47.45, the FAA Aircraft Registry must be notified within 30 days of any change to the aircraft registrant's address. Such notification is made by submitting a FAA Form 8050-1, Aircraft Registration Application, to AFS-750 in Oklahoma City, Oklahoma.
- 26. The pilot in command shall ensure all passengers carried onboard the aircraft are necessary for the purpose of the intended flight, and will list each passenger by name on an aircraft manifest.

These operating limitations are a part of a special airworthiness certificate issued: January 4, 2008

Date of Limitations Issuance: January 4, 2008

Date of Expiration: February 3, 2008

ssued Bv.

Emery P. Wiltse DAR-609001-CE



EXPERIMENTAL AIRWORTHINESS PROGRAM LETTER

1. Registered Owner Information (as shown on Certificate of Registration):												
Name	Address	Address										
Gulfstream Aerospace Corp.	•	500 Gulfstream Rd. MSC B-16 Savannah, GA 31408-9643										
2. Aircraft Description:												
Aircraft Registration	Aircraft Manufacturer	Year Mfg.										
N764GA	Gulfstream		2007									
Aircraft Serial No.	Aircraft Model											
5164 GV-SP (G550)												

3. Program Purpose

To flight test the aircraft under experimental airworthiness for the purposes of Research and Development and Showing Compliance to the Regulations for testing under STC Project ST6387CH-T, which will include flight operations for the forward cabin bulkhead electric pocket door and for fit and operation tests of cabin furnishings.

4. Estimated flight hours required: 30 hours

Estimated number of flights required: 8 flights

Estimated duration for program: 30 days

5. Flight Test Area Description (include address of base station)

The flight test area is defined as:

While at Appleton Wisconsin (ATW)

1000 NM radius around the Outagamie County Regional Airport, Appleton, Wisconsin (ATW), excluding Canadian airspace. Except for takeoff and landing operations this aircraft must be operated over open water or in sparsely populated areas having light air traffic.

6. Aircraft Configuration Description (attach drawings or photographs if necessary)

The aircraft will be configured with an aircraft interior accomplished under STC Project ST6387CH-T.

December 18, 2007

7. Date

Name & Title (type or print)

Carl C. Coonce, Manager, Final Phase Operations

Signature

· .

US Department of Transportation **MAJOR REPAIR AND ALTERATION**

Form Approved

OMB No. 2120-0020

For FAA Use Only

Federal Aviation

(Airframe, Powerplant, Propeller, or Appliance)

Office Identification

Administration														
INSTRUCTIO	NS: Print or typ	e all	entries. See FA	AR 43.9,	FAR	43 Appendix E	3, and AC	43.9-1 (or subsequ	uent revi	sion thereof) f	or instructions			
for each viola	on of this form. tion (Section 90	I NIS I	report is require	d by law	(49 1059	U.S.C. 1421). I	-allure to i	report can result in	a civil p	enalty not to e	exceed \$1,000			
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NOTICE

Weight and balance limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed).

Converted aircraft from production (green) aircraft to a Seventeen (17) Passenger Corporate Executive Cabin Interior in accordance with Gulfstream Aerospace Corp. **STC ST02424CH-D** issued 01/14/2008.

Structural modifications required to complete the Corporate Interior Installations are found on: Structures 8110-3 Drawing List for GV-SP S/N 5164 #GC520000238, Rev.--, which are DER approved by Mark G. Shallow, DERY-410138-CE on FAA 8110-3 form dated 01/14/2008.

Mechanical System Drawing 8110-3 List for GV-SP S/N 5164 #GC520000239, Rev.-- which are DER approved by Mark G. Shallow, DERY-410138-CE on FAA 8110-3 form dated 01/14/2008 and by Roger W. Duffrin DERY- 405049-CE on FAA 8110-3 form dated 01/14/2008.

The Completion Stress Report #GC524965164, Rev.-- for GV-SP S/N 5164 is DER approved by Mark G. Shallow, DERY-410138-CE on FAA 8110-3 form dated 01/14/2008.

The following Interior Drawings are DER approved by Emery P. Wiltse DERY-609001-CE on FAA 8110-3 form dated 01/14/2008.: GC52804K010, Rev.-- - Interior Compliance Checklist and GC520000226, Rev.-- - Compliance Data List, GV-SP S/N 5164.

Materials (including finishes or decorative surfaces applied to the materials) used in the installation of the new Corporate interior have been tested in accordance with Gulfstream Aerospace Burn Test Procedures Manual GACA-BT-001M Rev. 1 dated 5/00. Materials meet the flammability requirements of FAR 25.853 (a) Appendix F Part I (a) (1), FAR 25.855 (d) Appendix F Part I (a) (2). The galley and lavatory waste compartments meet the flammability requirements of FAR 25.853 (h).

Electronic Cable and additional wire used complies with the burn testing requirements of FAR 25.869 (a)(4) Appendix F Part I (a)(3) Amendment 25-113 which is DER approved by Carin Demus, DERY-405143-CE on FAA 8110-3 forms dated 11/21/2007.

Thermal/acoustic insulation materials installed in the fuselage meet the flame propagation burn tests of FAR 25.856 (a) Appendix F Part VI which are DER approved by Herb Reed II, DERT-230293-CE on FAA 8110-3 forms dated 10/13/2007.

All single and double cabin seating, excluding the jumpseat, which is certified from production, comply with the requirements of FAR 25.853 (c), Amd. 25-116, Appendix F Part II and FAR 25.853 (a), Amd. 25-116, Appendix F Part I (a)(1)(ii) in accordance with Skandia Inc., Test Plan #16998 Rev. A dated 12/18/2007 and Rev. IR dated 12/12/2007. DER approval is by Donna J. Parrish, DERY-410100-CE on FAA 8110-3 forms dated 12/18/2007.

Airplane Flight Manual Supplement, **GC52204M024** Rev.--, dated 01/14/2008 is required for this approval and is inserted in the applicable Airplane Flight Manual.

Instructions for Continued Airworthiness: Reference Gulfstream Aerospace Corp. Instructions for Continued Airworthiness Document No. **GC52801A480** dated 01/2007.

Attachment to Interior Completion 337 dated 01/15/2008 GV-SP (G550), S/N: 5164

1)

Installed Electric Window Shades (ATG) in accordance with Gulfstream Aerospace Corp. **STC ST03290AT-D** Issued 11/21/06.

Airplane Flight Manual Supplement No. **GC51506M002** dated 11/21/06 is required for this approval and is inserted into the appropriate Airplane Flight Manual.

Instructions for Continued Airworthiness: Reference Gulfstream Aerospace Corp Instructions for Continued Airworthiness Report No. **GC51 506A002** Rev.--, dated 11/15/2006.

Installed a Passenger Oxygen System (-103 config.) in accordance with Gulfstream Aerospace Corp. STC ST02696AT-D, Issued 08/19/2003.

Deviations are DER approved on Oxygen System STC #ST02696AT-D (-103 Config.) Deviation Drawing List #GC520000240 for GV-SP S/N 5164 Mark G. Shallow, DERY-410138-CE on FAA 8110-3 form dated 01/14/2008.and by Roger W. Duffrin DERY- 405049-CE on FAA 8110-3 form dated 01/14/2008.

Airplane Flight Manual Supplement, **GC51453M004** Rev. A dated 12/08/2004 is required for this approval and is inserted in the applicable Airplane Flight Manual.

Instructions for Continued Airworthiness: Reference GAC Instructions for Continued Airworthiness, Document Number: **GC51453A003** Rev. B dated 08/25/2004.

Installed Interior Emergency Lights in accordance with Gulfstream Aerospace Corp. STC ST02697AT-D, Issued 08/20/2003.

Instructions for Continued Airworthiness: Reference GAC Instructions for Continued Airworthiness, Report Number: **GC51486A002**, Rev. A dated 02/12/2004.

Installed an External Camera System (Securaplane) in accordance with Gulfstream Aerospace STC ST02715AT-D issued 10/07/2003.

Airplane Flight Manual Supplement, GC51505M006 Rev. A dated 09/21/2004 is required for this approval and is inserted in the applicable Airplane Flight Manual.

Instructions for Continued Airworthiness: Reference Gulfstream Aerospace Corp. Instructions for Continued Airworthiness Document No. **GC51505A005** Rev. B dated 09/03/2004.

Installed a 115VAC/60 Hz Cabin Outlets System in accordance with Gulfstream Aerospace STC ST02292CH-D issued 11/17/2006.

Airplane Flight Manual Supplement, GC51 494M000 Rev.-- dated 11/17/2006 is required for this approval and is inserted in the applicable Airplane Flight Manual.

Instructions for Continued Airworthiness: Reference Gulfstream Aerospace Corp. Instructions for Continued Airworthiness Report No. **GC51494A000** Rev.-- dated 10/31/2006.

Installed an Airshow 4000 System in accordance with Gulfstream Aerospace STC ST02048CH-D issued 06/01/2005.

Airplane Flight Manual Supplement, GC51504M000 Rev.- dated 06/01/2005 is required for this approval and is inserted in the applicable Airplane Flight Manual.

Instructions for Continued Airworthiness: Reference Gulfstream Aerospace Corp. Instructions for Continued Airworthiness Document No. **GC51504A001** Rev. A dated 03/15/07.

Installed a Broad Band Multi-Link (BBML) System in accordance with Gulfstream Aerospace STC ST02796AT-D issued 01/21/05. last amended 10/26/2006.

Airplane Flight Manual Supplement, **GC51303M005** dated 05/08/2006 is required for this approval and is inserted in the applicable Airplane Flight Manual.

Instructions for Continued Airworthiness: Reference Gulfstream Aerospace Corp. Instructions for Continued Airworthiness Document No. **GC51303A006** Rev.-- dated 06/27/2006.

Installed a Securaplane 500 Security System in accordance with Gulfstream Aerospace STC ST02049CH-D issued 06/01/2005.

Instructions for Continued Airworthiness: Reference Gulfstream Aerospace Corp. Instructions for Continued Airworthiness Document No. **GC51456A000** Rev. A dated 03/15/2007.

Installed a Airshow RF/IF Remote Controller with Touchscreen Remote System to the Airshow Cabin Management System in accordance with Gulfstream Aerospace STC ST01957CH-D issued 09/21/04.

Instructions for Continued Airworthiness: Reference Gulfstream Aerospace Corp. Instructions for Continued Airworthiness Document No. **GC51602A000** Rev.-- dated 09/21/2004.

Installed Emergency Vision Assurance System (EVAS) in accordance with Vision Safe Corporation STC ST00892LA issued 12/15/1999.

Airplane Flight Manual Supplement **No. 1** Rev. 1 dated 11/08/2004 is required for this approval and is inserted into the appropriate Airplane Flight Manual.

Instructions for Continued Airworthiness: Reference EVAS ICA Document Number: 8017, Rev. 5 dated 02/12/2007 for Maintenance Requirements.

Installed **ASC 049** dated 05/02/2006 - Navigation (ATA 34) Lightning Sensor System (LSS) Installation using Gulfstream Aerospace **STC ST02915AT-D** issued 05/122005.

Airplane Flight Manual Supplement, **GC51 225M000** dated 05/12/2005 is required for this approval and is inserted in the applicable Airplane Flight Manual.

Instructions for Continued Airworthiness: Reference Gulfstream Aerospace Corp. Instructions for Continued Airworthiness Document No. **GC51 225A000** Rev.-- dated 04/05/2005.

DRAWING	SYSTEM
GC51 1070307, Rev	Call System
GC51 1790268, Rev	Cockpit Lights (Mod)
GC51 3700325, Rev	Cockpit Switch Panels
GC51 4620102, Rev	Rechargeable Flashlights (Mod)
GC51 4780310, Rev	Passenger Warning Signs
GC51 4990233, Rev	Spare Wires
GC51 6190204, Rev	Main Cabin Door Switch Relocation
GC51 5900239, Rev	Galley Lights
GC51 4530198, Rev	Oxygen System (Mod)
GC51 4860357, Rev	Interior Emergency Lights
GC51 4864149, Rev	Wire Routing Diagram Interior Emergency Lights
GC51 5410190, Rev	Galley Oven
GC51 5420031, Rev	Hot Cup
GC51 5430195, Rev	Coffee Maker
GC51 5460292, Rev	Galley Outlets
GC51 5480319, Rev	Water System (Mod)
GC51 6960296, Rev	Aft Lav Électrical
GC51 7460177, Rev	Thermal Electric Cooling Unit
GC51 4160341, Rev	Cabin Switch Block Diagram
GC51 5690329, Rev	Cabin Data Bus
GC51 5710257, Rev	VIP/Entertainment Switch Panels
GC51 5730250, Rev	Galley Switch Panels
GC51 3910011, Rev	R EER CB Panel (Mod)
GC51 3940345, Rev	Aux CB Panel (Mod)
GC51 5400308, Rev	Galley CB Panel
GC51 3034046, Rev	Wire Routing Diagram Radio Satcom HSD (Skylink)
GC51 4284102, Rev	Wire Routing Diagram Wireless LAN (BBML)
GC51 4294028, Rev	Wire Routing Diagram LAN Server (BBML)
GC51 4304014, Rev	Wire Routing Diagram Voice Over IP (BBML)
GC51 4564018, Rev	Wire Routing Diagram Security System
GC51 4934006, Rev	Wire Routing Diagram 60 Hz Converter
GC51 4944017, Rev	Wire Routing Diagram 60 Hz Outlets
GC51 4980134, Rev	Wire Routing
GC51 5054145, Rev	Wire Routing Diagram Camera System
GC51 5064139, Rev	Wire Routing Diagram Electric Window Shades
GC51 5540048, Rev	LH Galley Assembly
GC51 6540125, Rev	R/H Galley Annex Assembly
GC51 6940121, Rev	Aft LH Vanity Assembly

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REPORT

SYSTEM

GC51 4696353, Rev.--

Electrical Load Analysis Supplement

Electrical / electronic aspects of the above listed electrical drawings are DER approved by Roger W. Duffrin DERY- 405049-CE on FAA 8110-3 forms dated 01/14/2008.

DRAWING SYSTEM GC51 1030299, Rev.--Satcom (Mod) GC51 1040322, Rev .--Radio Telephone (System Removal) GC51 1090297, Rev.--Cockpit Audio (Mod) GC51 3240350, Rev.--Ident Strapping GC51 1650117, Rev.--Forward Cabin Disconnects Aft Cabin Disconnects GC51 2650117, Rev.--GC51 3650132, Rev.-Miscellaneous Disconnects **Aux Avionics Cooling** GC51 1990222, Rev.--GC51 4230085, Rev.--ELT (MOD) GC51 5760195, Rev.--Valance Panel Disconnects GC51 6460090, Rev.--Crew Galley/Refreshment GC51 5930042, Rev.--FWD Crew Refreshment Assembly GC51 2030054, Rev.--Aircell Iridium Satcom (Axxess II) GC51 3030087, Rev.--Satcom High Speed Data (Skylink) GC51 4280202, Rev--Wireless LAN (BBML) GC51 4290057, Rev--LAN Server (BBML) GC51 4300013, Rev--Voice Over IP (BBML) GC51 4560199, Rev.--Security System (Securaplane) GC51 4630260, Rev.--Nose External Switch Panel (Mod) 60HZ Converter (Mod) GC51 4930225, Rev.--GC51 4940274, Rev.--60HZ Outlets

Electrical/electronic aspects of the above listed electrical drawings are DER approved by Calvin A Lewis DERY 405115-CE on FAA 8110-3 forms dated 01/09/2008.

DRAWING SYSTEM GC51 4520187, Rev.--Cabin Temp Sensor Relocation GC51 4570294, Rev.--Cabin Temp Control (Mod) GC51 4820307, Rev.--Vestibule Electrical GC51 5060177, Rev.--**Electric Window Shades** GC51 5810309, Rev.--Cabin Effect Lights GC51 5820317, Rev.--Cabin Spot Lights GC51 5840301, Rev.--Aisle Lights GC51 5870275 Rev.--Closet Lights GC51 5940098, Rev.--Fwd Lavatory Assembly GC51 5960288, Rev.--Fwd Lavatory Electrical GC51 6880086, Rev.--Aft LH Closet Assembly GC51 7880113, Rev.--Aft RH Closet Assembly

Electrical / electronic aspects of the above listed electrical drawings are DER approved by Emery P. Wiltse DERY-609001-CE on FAA 8110-3 form dated 01/09/2008.

DRAWING SYSTEM GC51 4270277, Rev.-Fax Interface GC51 5010319, Rev.--Cabin Audio GC51 5020344, Rev.--Cabin Entertainment GC51 6010257, Rev.--Cabin Headphones GC51 7020133, Rev.--Credenza Assembly GC51 5050245, Rev .--Camera System (AVS) GC51 5050244, Rev.--Camera System (Securaplane) GC51 6020182, Rev.--Remote Control (RF) GC51 5040316, Rev.--Cabin Display (Airshow 4000)

Electrical/electronic aspects of the above listed electrical drawings are DER approved by Peter Jobe DERY 830436-CE on FAA 8110-3 forms dated 01/11/2008.

Aircraft wiring diagrams affected by the above installations are revised to reflect changes in system design or their interface to newly installed or modified systems as appropriate. All new single conductor wire meets MIL22759/16 and all new multi-conductor shielded wire meets MIL 27500.

Reference GAC serialized Outfitters Equipment List and / or electrical drawings bill of materials for details on equipment installed in conjunction with this installation. Aircraft was re-weighed.

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UNITED STATES OF AMERICA DEPARTMENT OF TRANSPORTATION—FEDERAL AVIATION ADMINISTRATION

STANDARD AIRWORTHINESS CERTIFICATE

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1.	NATIONALITY AND REGISTRATION MARKS	2. MANUFACTURER AND MODEL	3	AIRCRAFT SERIAL NUMBER	4 CATEGORY
	₩372BG	Gulfstream GV-SP (G550)		5164	Transport
5	AUTHORITY AND BAS	IS FOR ISSUANCE			

This airworthiness certificate is issued pursuant to the Federal Aviation Act of 1958 and certifies that, as of the date of issuance, the aircraft to which issued has been inspected and found to conform to the type certificate therefor, to be in condition for safe operation, and has been shown to meet the requirements of the applicable comprehensive and detailed airworthiness code as provided by Annex 8 to the Convention on International Civil Aviation, except as noted herein,

Exceptions: 7946 25.813(e) Door Between Passenger Compartments 8004 25.901(c) Single Failure Criteria Exemption Exemption Exemption 8142 25.901(c) Single Failure Criteria

6. TERMS AND CONDITIONS

Unless sooner surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator, this airworthiness certificate is effective as long as the maintenance, preventative maintenance, and alterations are performed in accordance with Parts 21, 43, and 91 of the Federal Aviation Regulations, as appropriate, and the aircraft is registered in the United States.

DATE OF ISSUANCE

R 8/21/07

FAA REPRESENTATIVE

Emery P.

DESIGNATION NUMBER

DART-609001-GL

Any alteration, reproduction, or misuse of this certificate may be punishable by a fine not exceeding \$1,000, or imprisonment not exceeding 3 years, or both. THIS CERTIFICATE MUST BE DISPLAYED IN THE AIRCRAFT IN ACCORDANCE WITH APPLICABLE FEDERAL AVIATION REGULATIONS



UNITED STATES OF AMERICA DEPARTMENT OF TRANSPORTATION—FEDERAL AVIATION ADMINISTRATION

STANDARD AIRWORTHINESS CERTIFICATE

۱.	NATIONALITY AND REGISTRATION MARKS
	[*] N764GA

2. MANUFACTURER AND MODEL

Gulfstream GV-SP(G550)

AIRCRAFT SERIAL NUMBER 5164

4. CATEGORY

Transport

5. AUTHORITY AND BASIS FOR ISSUANCE.

This airworthiness certificate is assued pursuant to the Federal Aviation Act of 1958 and certifies that, as of the date of issuance, the aircraft to which issued has been inspecified and found to conform to the type certificate therefor, to be in condition for safe operation, and has been shown to meet the equipments of the papplicable comprehensive and detailed airworthiness code as provided by Annex 8 to the Convertion on International Civil Aviation except as noted herein.

Exemption No. 7946/25.813(E) Door Between Passenger Compartments
Exemption No. 8004-25.901.(C) Single Failure Criteria

Exemption No. 8142-25.901(c) Single Failure Criteria

6. TERMS AND CONDITIONS

Unless sooner surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator, this airworthiness certificate is effective as long as the maintenance preventative maintenance, and alterations are performed in accordance with Parts 21, 43 and 91 of the Federal Aviation Repulations, as appropriate and the aircraft is registered in the United States

DATE OF ISSUANCE

FAA RERRESENTATIVE

Jessie K. Jones

DESIGNATION NUMBER

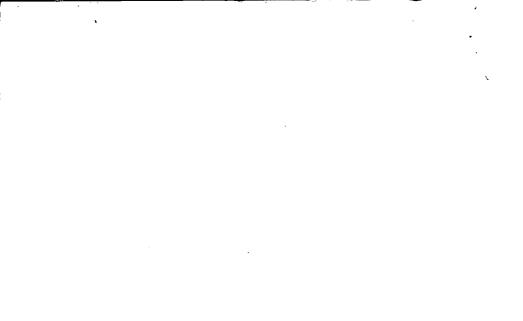
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Any alteration, reproduction, or misuse of this certificate may be punishable by aline not exceeding \$1,000, or imprisonment not exceeding 3 years, or both. THIS CERTIFICATE MUST BE DISPLAYED IN THE AIRCRAFT IN ACCORDANCE WITH APPLICABLE FEDERAL AVIATION REGULATIONS.

FAA Form 8100-2 (8-82)

A08/21/2007

U.S. GPO-2001 - 668-455



FAA FORM 8130-6, APPLICATION FOR U.S. AIRWORTHINESS CERTIFICATE

Form Approved

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	C 6 MULTIPLE AIRWORTHINESS CERTIFICATE (check ABOVE "Restri								cted Opera	tion" ar	nd "Standar	d" or "Li	mited" a	as applic	cable)	_			
	A. REGISTERED OWNER (As shown on certificate of aircraft registration)											EALER,						+ ✓	
	NAME Gul	fstre	am .	Aerospace C	orporation					ADD	DRESS 5(00 Gi	ılfstr	eam	Rd., Sav	vannah, G	A 3	1408-9643	
	Gulfstream Aerospace Corporation B. AIRCRAFT CERTIFICATION BASIS (Check applicable blocks and complete items as																		-
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CA	AIRCRAFT SPECIFICATION OR TYPE CERTIFICATE DATA SHEET (Give Revision No.) A12EA Rev. 29									number of the last AD SUPPLEMENT available in the biweekly series as of the date of application)									
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_	accorda	nce v	vith Ti	tle 49 of the Unit	ed States Code 4	4101_6	t sec	and applicable	Federal	Aviati	on Regula	ations,	and th	at the	aircraft has	s been inspe	cted	and is airworthy and eligible	for the
	DATE C				NAME AND	TITLE	E (Prin	nt or type)				_	Ts	IGNAT	TÜŘE				
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IV. INSPECTION AGENCY VERIFICATION								· · · · · · · · · · · · · · · · · · ·											
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<u>₩</u> ≥				-	Section I or IV m	eets re	quire	ments for		4					REQUESTE			MINORELINESS CERTIC	ICATE
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DDUC													,	
VI. PRODUCTION FLIGHT TESTING			RTIFICATE C	FICATE (Give production)	on certifica	te number) —				 				
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ŀ	C. G	VE QUANTITY O	F CERTIFICA	TES REQUIRED FO	OR OPER	RATING NEEDS								
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	A. DI	SCRIPTION OF	AIRCRAFT	<u> </u>										
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	BUILI	DER (Meke)			_		MODEL							
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	B. D	SCRIPTION OF	FLIGHT			CUSTOMER DE	MONSTRA	TION	LIGHTS [(Check if app	olicable)			
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	F. Cl accor	ERTIFICATION – dance with Title 4	I hereby certif 19 of the United	y that I am the regist d States Code 44101	ered own 1 <u>et seq.</u>	ner (or his agent) of the and applicable Feder	the aircraft described above; that the aircraft is registered with the Federal Aviation Administration in eral Aviation Regulations; and that the aircraft has been inspected and is safe for the flight described.							
	DATE		NAME .	AND TITLE (Print or T	уре)		SIGNATURE							
9	✓	A. Operating Li as applicabl		Markings in Compliar	nce with	14 CFR Section 91.9	·	G. Statement of Conformity, FAA Form 8130-9 (Attach when required)						
SSS		B. Current Ope	rating Limitatio	imitations Attached					oreign Airwor Attach when		ation for Import Aircraft			
RTHINE AA/DES		C. Data, Drawin	ngs, Photograp	hs, etc. (Attach when r	required)			I. P	I. Previous Airworthiness Certificate Issued in Accordance with 14 CFR Section 21.197(a) (3) CAR (Original Attached) J. Current Airworthiness Certificate Issued in Accordance with 14 CFR Section 21.183 (a) (Copy Attached)					
AIRWO TION (F.	1	D. Current Wei	ght and Baland	ce information Availa	ble in Air	rcraft		14 0				(Original Att	ached)	
VIII, AIRWORTHINESS DOCUMENTATION (FAADESIGNEE use only)		E. Major Repai	r and Alteration	n, FAA Form 337 (Am	tach when	required)	✓	J. 0				hed)		
) 0CF		F. This inspection Recorded in Aircraft Records						К. І	ight-Sport Air	craft Statement	of Compliance, FAA Form	8130-15 (Attach	when	

FAA Form 8130-6 (10-04) Previous Edition Dated 5/01 May be Used Until Depleted, except for Light-Sport Aircraft

NSN: 0052-00-024-7006

UNITED STATES OF AMERICA
DEPARTMENT OF TRANSPORTATION—FEDERAL AVIATION ADMINISTRATION
STANDARD AIRWORTHINESS CERTIFICATE

NATIONALITY AND REGISTRATION MARKS 2. MANUFACTURER AND MODE AIRCRAFT SERIAL 4. CATEGORY **N764GA Transport** This airworthiness certificate is lauded pursuant of the leader of the date of issuance, the aircraft to which issued has been inspecied and found to conform to this type certificate therefor, to be in condition for safe operation, and has been shown to meet the requirements of the applicable comprehensive and detailed airworthiness code as provided by Annex 8 to the Convention on International critical provided by Annex 8 to the Convention on International critical provided by Annex 8 to the Convention on International critical provided by Annex 8 to the Convention on International critical provided by Annex 8 to the Convention on International critical provided by Annex 8 to the Convention No. 7946 25:813 (c) Door Between Passenger Compartments Exemption No. 8004-75:901 (c) Single Failure Criteria Exemption No. 8142-25.901 (c) Single Failure Criteria S AND CONDITIONS Unless sooner surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator, this airworthiness certificate is effective as long as the maintenance preventative maintenance, and alterations are performed in accordance with Parts 21, 43 and 9101 the Federal antition Regulations, as appropriate and the aircraft is registered in the United States. 6. TERMS AND CONDITIONS FAA REPRESENTATIVE DATE OF ISSUANCE DESIGNATION NUMBER Jessie K. Jones 08/20/2007 ODARF100127CE Any alteration, reproduction, or misuse of this certificate may be punishable: by a line not exceeding \$1,000, or imprisonment not exceeding \$3 years, or both. THIS CERTIFICATE MUST BE DISPLAYED IN THE AIRCRAFT IN ACCORDANCE WITH APPLICABLE FEDERAL AVIATION REGULATIONS.

U.S. GPO-2001 - 668-455

FAA Form 8100-2 (8-82)

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FAA FORM 8130-6, APPLICATION FOR U.S. AIRWORTHINESS CERTIFICATE

Form Approved

<u> </u>																		O.M.B. No. 2120	-0018
U.S. Dep of Transp Federal Adminis	oortation Aviation tration			U.S. (PLICATION AIRWOR	TI	IIN TE	ESS	o. re	nly. Sul equired, /II as ap	bm us pli	it original o se attachm cable.	only t ent.	o ar Foi	n authorized r special fli	d FAA Repright permits	esen s co	areas; these are for FAA ntative. If additional spac omplete Sections II, VI	ce is
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1. AIRCRAFT DESIGNATION	N764GA Gulfstream A 5. AIRCRAFT SERIAL NO. 6. ENGINE BUILD											V-SP (G5 DEL DESIGN				2007	إΩ	MOD APS	2
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		9	H	LIGHT-SPORT	(Indicate Class)			AIRPLANE	PO	WER-PAR	ACH	RUTE	WEIG	HT-SI	HIFT-CONTROL	GLIDI	ER	LIGHTER THAN AIR	
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_		5		PROVISIONAL	(Indicate Class)	2		CLASS II											
CERTIFICATION REQUESTED						1	T	AGRICULTUR	E AN	D PEST (CO	NTROL	2	7	AERIAL SUR	VEY	3	AERIAL ADVERTISING	
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CA			,			4	+-	AIR RACING					5		CREW TRAIN		6	MARKET SURVEY	
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						+	╅╴	FERRY FLIGHT FOR REPAIRS, ALTERATIONS, MAINTENANCE, OR STORAGE EVACUATION FROM AREA OF IMPENDING DANGER								ory an worth mess certaicate under 9 :	21.190		
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				applicable on rever		4		DELIVERING (XPORTIN	IG		5	Ţ	PRODUCTIO	N FLIGHT TE	STIN	NG		
								CUSTOMER D	EMC	EMONSTRATION FLIGHTS .									
	C 6	<u> </u>	MU	LTIPLE AIRWOR	RTHINESS CER	TIFI	CATE	(check ABOVE "Res	stricted	Operation	an	d "Standard" or '	"Limite	d" as	applicable)				
	A. REGISTERED OWNER (As shown on certificate of aircraft registration)										ADDRESS 500 Gulfstream Rd., Savannah, GA 31408-9643								
	Gulfstream Aerospace Corporation									^'	טט	^{KESS} 500 C	Gulf	stre	am Rd., S	avannah, C	GA :	31408-9643	
7	B. AIRCRAFT CERTIFICATION BASIS (Check applicable blocks and complete items a									cated)							-		
ERTIFICATION	AIRCRAFT SPECIFICATION OR TYPE CERTIFICATE DATA SHEET (Giv								ne No.	e No. and AIRWORTHINESS DIRECTIVES (Check if all applicable AD's are compiled with and give the number of the last AQ SUPPLEMENT available in the biweekly series as of the date of application) 2007-1 7-01									
5				A12EA F STING (Give page						▼ 2007-17-01 SUPPLEMENTAL TYPE CERTIFICATE (List number of each STC incorporated)									
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Ħ	accorda	nce v	vith Ti															e Federal Aviation Administral d and is airworthy and eligible	
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V. FAA REPRESENTATIVE CERTIFICATION	B. Inspe			ahanai hatuit fil	Mei GeciiOU VII	r/dS		CERTI		TE HOLD)E'F	NUNDER	Н		DESIGNEE OFR part 65	14 CF	R par	л 121 OR 135 14 CFR р	part 145
V. RESI	DATE			DISTRICT OF	FICE	\neg	DES	GNZE'S SIGNAT					بــا			NSPECTO	30	ZEIVED ATLAN	
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FAA Form 8130-6 (10-04) Previous Edition Dated 5/01 May be Used Until Depleted, Except for Light-Sport Aircraft

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VI. PRODUCTION FLIGHT TESTING	A. M.	ANUFAC	TURER											
	NAME						ADDRESS							
	В. Р	RODUC	TION BASIS	Check applic	able item)	•								
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	A. D	ESCRIP	TION OF A	RCRAFT										
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	SER	IAL NUN	BER				REGIST	RATION MARK	c					
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	accordance with Title 49 of the United States Code 44101 et seq. and applicable Feder DATE NAME AND TITLE (Print or Type)								SIGNATURE					
953	✓		erating Limi applicable.	tations and M	Markings in Complia	nce with 14 CFR Section 9	91.9,	G. Statement of Co.	nformity, FAA	Form 8130-9 (Attach when	required)			
GNEE		B. Cu	rrent Operat	ling Limitatio	ns Attached			H. Foreign Airworth (Attach when red						
THINE:	C. Data, Drawings, Photog				hs, etc. (Attach when	required)	,			ite Issued in Accordance wit	h			
AIRWOI TION (F)	✓	D. Cu	rrent Weigh	t and Balanc	e information Availa	ble in Aircraft	-	14 CFR Section 2	1.183 (a)	CAR	(Original Attached)			
VIII. AIRWORTHINESS DOCUMENTATION (FAA/DESIGNEE use only)		E. Ma	jor Repair a	nd Alteration	, FAA Form 337 (At	ach when required)	✓	J. Current Airworthin	ness Certificat 21.183 (d)	e Issued in Accordance with	1 (Copy Attached)			
DOCE	1	F. Thi	s inspection	Recorded in	Aircraft Records	٠,, , .				of Compliance, FAA Form 8				

DEPARTM	ENT OF TRANSPORTATION—FE	DERAL AVIATION ADMIN	ISTRATION -
STA	NDARD AIRWORTH	INESS CERTIFIC	ATE ,
NATIONALITY AND 2. REGISTRATION MARKS	MANUFACTURER AND MODEL	3. AIRCRAFT S NUMBER	ERIAL 4. CATEGORY
N764GA	Gulfstream GV-SP	5164	Transport
AUTHORITY AND BASIS FO	OR ISSUANCE V		
operation, and has be provided by Annex 8 to	pertificate is basued pursuant to the li- to which issued has been in specification to een shown to meet the requirements of the Convention on International Givil Avia emption No. 7946/25:813 (c) Do Exemption No. 8004-25:901-(c)	ing applicable comprehensive ion except as noted herein. or Between Passenger Co Single Pailure Criteria	and detailed airworthiness code as
·	Exemption No. 8142.25.901(c)	Single Failure Criteria	
States.	indered suspended, revoked, or a lenate is effective as long as the maintent is 21, 43, and a for the Federal Avaliance.	mination date is otherwise esta ance preventative mainlenance sgulations, as appropriate and the	ablished by the Administrator, this, and alterations are performed in the united
ATE OF ISSUANCE A08/21/2007	FAA REPRESENTATIVE	ni fins	DESIGNATION NUMBER
	isuse of this certificate may be punishable ATE MUST BE DISPLAYED IN THE	K. Jones	ODARF100127CE
AA Form 8100-2 (8-	82)		U.S. GPO-2001 - 668-455
	UNITED STATES MENT OF TRANSPORTATION—F ANDARD AIRWORTH	EDERAL AVIATION ADMI	
. NATIONALITY AND 2. REGISTRATION MARKS		3. AIRCRAFT NUMBER	
N764GA	Gulfstream GV-SP	510	Transport
5. AUTHORITY AND BASIS This airworthiness issuance, the aircr operation, and has provided by Annex 8 Exceptions:	certificate is assued pursuant to the art to which issued has been in posting and been shown to meet the requirements to the Convention on International Civil Sur Exemption No. 7946 25:813 (Exemption No. 8004-25.80)	Sime Failure riteria	and certifies that, as of the date icate therefor, to be in condition for see and detailed airworthiness code r Compartments
TERMS AND CONDITION Unless sooner su airworthiness certii accordance with Pa States.	rendered: av Gode Toylor or rendered: av Gode Toylor or rendered: av Gode Toylor or rendered av Aviation	emination where is the was leen was prevently chaintenant the service of the serv	stablished by the Administrator, to ce, and alterations are performed to the aircraft is registered in the Unit
DATE OF ISSUANCE 08/29/2007	PAA HERRESEN I	K.Jones	DESIGNATION NUMBER ODARF100127CE
Any alteration, reproduction, or	misuse of this certificate may be punished FICATE MUST BE DISPLAYED IN TH	ble:by-a-fine not exceeding \$1,0	00, or imprisonment not exceeding
FAA Form 8100-2	(8-82)		U.S. GPO-2001 - 668-45

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